



Traveler Phase 1A Joint Review

6/20/2017



Preface



- Initial system design, implementation, verification, and flight testing has been conducted. As of yet detailed data review is incomplete, and flight testing has been limited to initial monitor force fights. Detailed monitor flight evaluations have yet to be performed. Initial preliminary findings are being discussed in this package. They are not final.
- Lessons learned guidance is preliminary as well and attempts have been made to tailor them to generic Multi-Monitor Run-Time Assurance (MMRTA). Guides are expected to be developed once more detailed testing / analysis has been performed.



Outline



- Expandable Variable Autonomous Architecture (EVAA) Phase 1A Description
 - Phase 1A Scope
 - Hardware
 - Software
- Findings from EVAA Phase 1A
 - Requirements Development
 - Design Development
 - Test Approach
 - Verification Testing
 - Flight Test
- Standard Practice Review
 - Review of ASTM standard practice
 - Considerations for autonomous systems certification framework / approach
- Next Steps
 - Now to 31 Aug 17
 - Potential Later Phases



EVAA Phase 1A Description



Phase 1 / 1A Scope



- Phase 1, Oct 2016-Sept 2017, Develop and evaluate an MMRTA system **utilizing memory/database of terrain & obstacles (e.g., not onboard sensors)**
 - Low risk, rapid prototyping approach for Phase 1 drove low cost, ~small sUAS
 - Resulted in light payload capability (<3 lbs), no onboard sensor capability
 - Demonstration of EVAA / MMRTA concept
- Phase 1A: Complete at least 1 flight test of a MMRTA test case involving command conflicts (i.e., simultaneous safety monitor/recovery controller command requests)
 - ~June 2017: Provide our findings to the ASTM WK53403 Standards Practice Report: Methods to Safely Bound Flight Behavior of UAS Containing Adaptive Algorithms
 - Assess performance of the Moral Compass
- Phase 1B: Evaluate Monitor/Recovery and Moral Compass performance
 - ~Sept 2017: Provide our findings to the ASTM WK53403 Standards Practice Report: Requirements Guide, Evaluation Guide, & Design Guide
- Phase II: (~FY18) Add Onboard sensors, aircraft detect & avoid, person avoidance
 - Larger aircraft, ~9-10 lbs payload capability, >30 min endurance



First Flight Day, 3 May 2017, Mission 23



- Objective:
 - Conduct Phase 1A testing to verify EVAA functionality for first time and with goal to assess Moral Compass “force fight”.
- Result:
 - Met objective (tested GCAS and GeoFence functionalities including a Moral Compass force fight).
 - Conducted ~17 runs for GCAS testing (included repeats due to team learning best method to setup the loiter portion of runs and transitions to the test profile run-ins, and assessing unexpected GCAS avoidances).
 - Conducted 2 runs of GeoFence – demonstrating its functionality.
 - Conducted 2 runs of a Multi-Monitor test profile: obstacle avoidance (GCAS) versus airspace avoidance (GeoFence) Moral Compass force fight.
 - GPS jamming being conducted in the Utah range may have influenced some of our data runs sporadically.



Second Flight Day, 18 May 2017, Mission 24



- Objective:
 - Conduct Phase 1A testing to verify EVAA functionality in an environment without GPS jamming (1st flight had GPS jamming in Utah, which did appear to affect our data) and following software fixes made to address discrepancies from EVAA flight #1 that occurred on 3 May.
- Result:
 - Met objective, flew all cards, including multi-monitor fights with GCAS, GeoFence, and Forced Landing System.
 - 26 test profiles conducted.
 - A few anomalies noted with GCAS (recovers 20-30 ft lower altitude than expected and appears to avoid ~ 2 sec later than expected). Team surmised there could be a difference between the onboard map location vs where aircraft thinks it is vs obstacle locations placed via Google Earth.

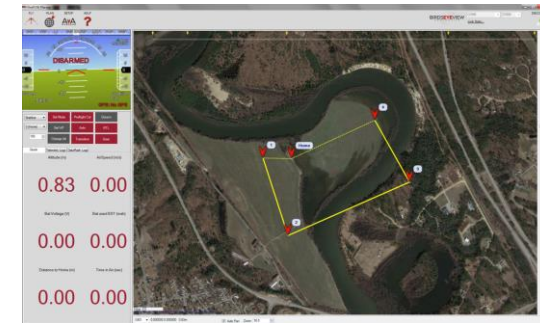
Note: Demonstrated system worked, met Objective of Phase 1A: “Complete at least 1 flight test of a MMRTA test case involving command conflicts”



Baseline Aircraft



- BirdsEyeView (BEV) Firefly 6
 - Cost: \$5,999.00 (fully assembled by manufacturer)
 - ~\$2500 (if you build it yourself)
- **Hardware**
 - COTS FireFLY6 by BEV Aerobotics
 - COTS Pixhawk autopilot hardware with BEV proprietary software
 - COTS FireFLY6 Planner Ground Station Software
 - COTS RC Controller
 - ODROID XU4 (EVAA is hosted on the ODROID)
 - USB connection to Pixhawk for autopilot interface



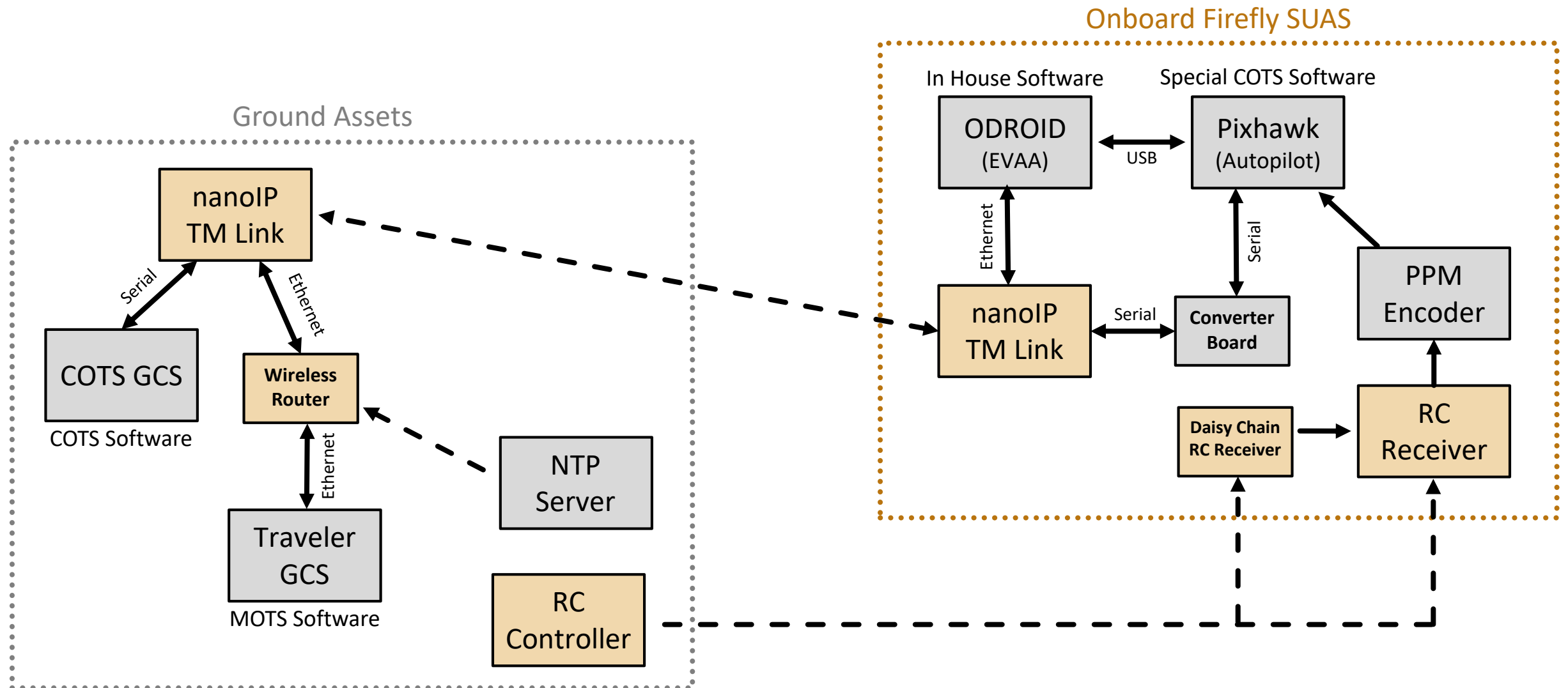
– **Features & Specifications**

- Vertical takeoff and landing with fixed wing flight mode
- Flight Modes – Stabilize, Alt Hold, Loiter, Land, Return-To-Launch, Waypoint
- Effectively constant airspeed of 30-35kts
- Max Vertical Velocity of ~ +/-10fps
- Max Bank Angle of +/- 40deg



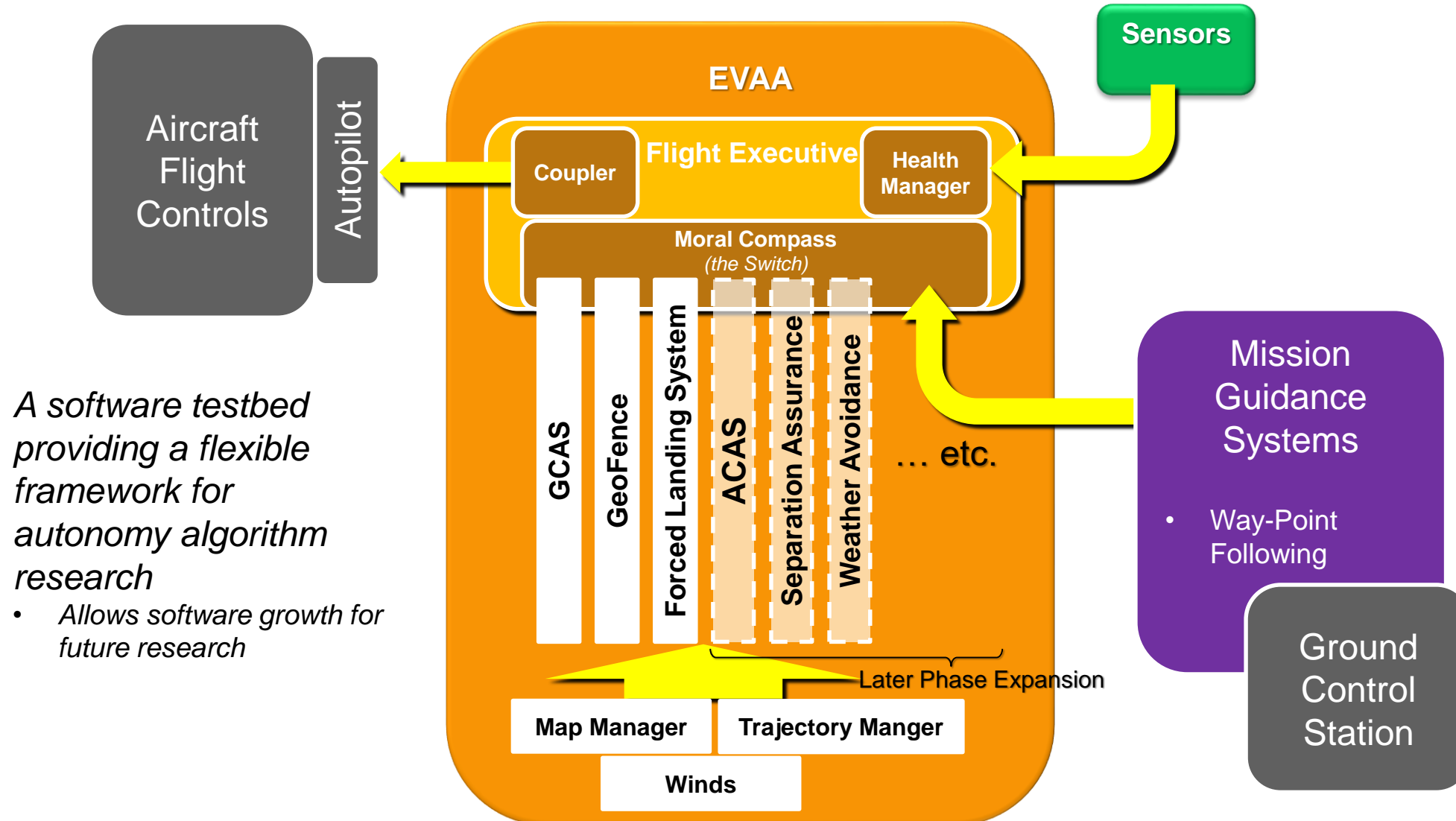


Simplified Traveler System Hardware Architecture





EVAA Software System Description

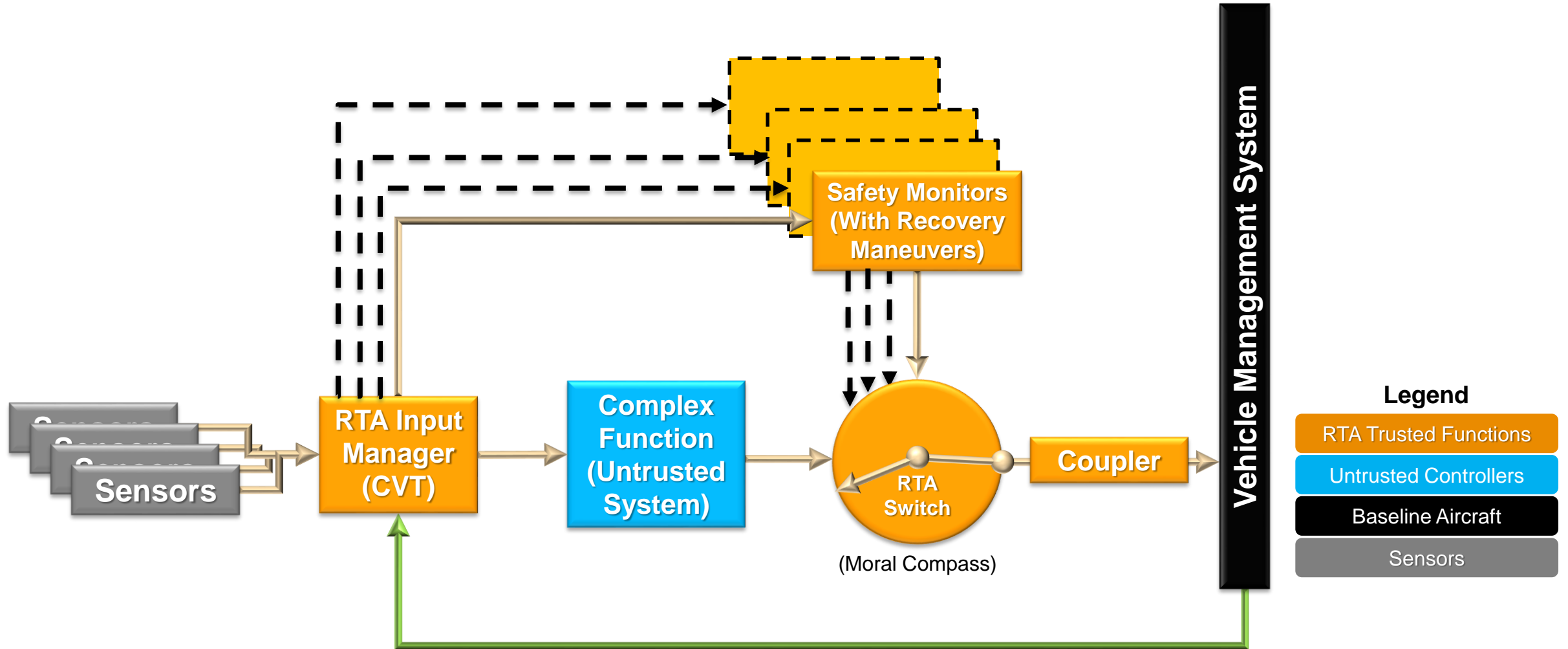


A software testbed providing a flexible framework for autonomy algorithm research

- Allows software growth for future research



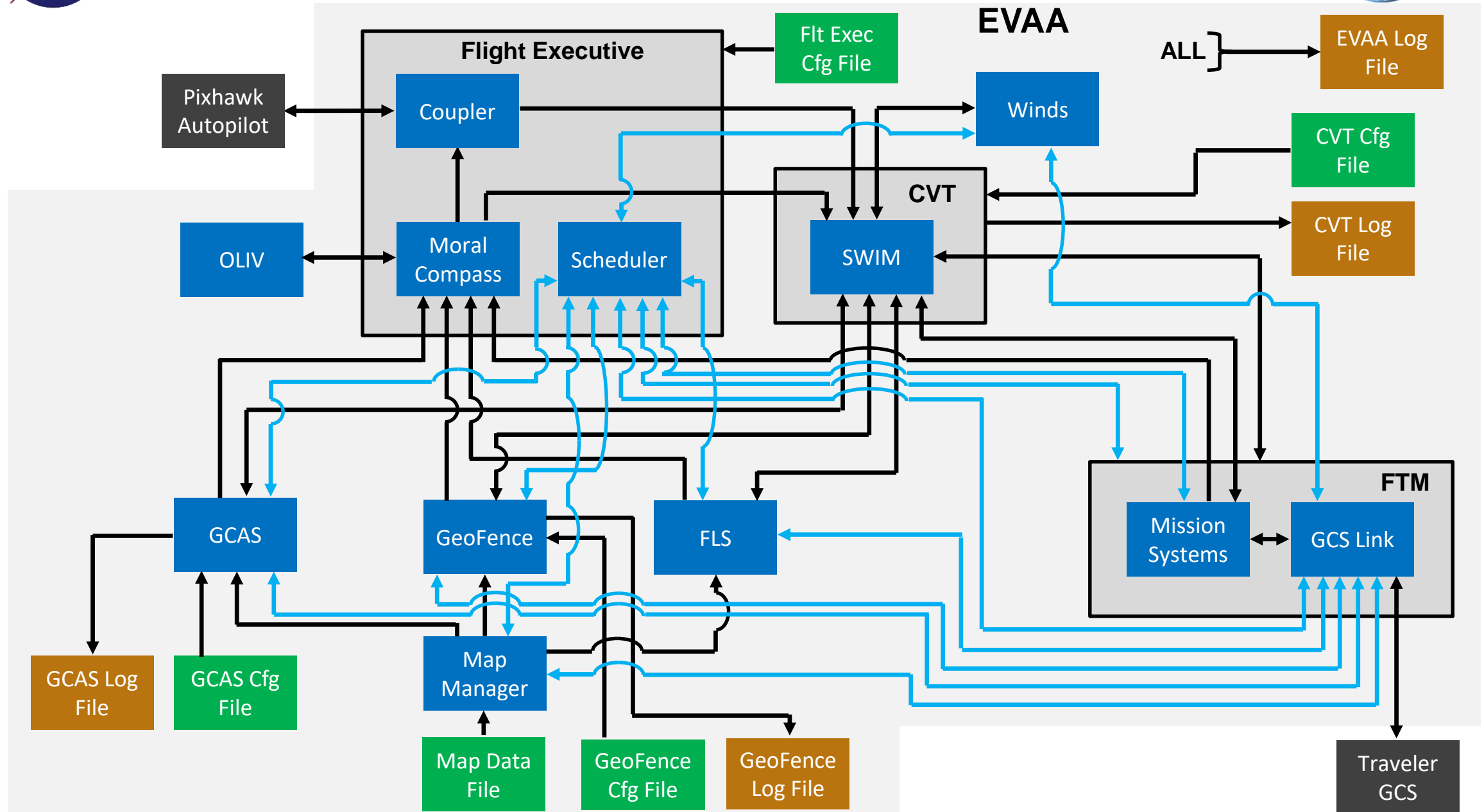
Current EVAA MMRTA Framework



*Note: Recovery maneuvers are contained within monitors, but may be updated via configuration file for modularity / adaptability across platforms.



Data Flow Interfaces





EVAA Software Description



- Flight Executive

- **Moral Compass**: Chooses the safety/recovery monitor with highest consequence command from all of the safety monitors to pass to the coupler. (Consequence: estimated non-dimensionalized monetary penalty if requested monitor action not acted upon)
 - **OLIV**: Checks to ensure valid command and consequences reported by the safety monitors
- **Scheduler**: Creates and schedules threads for all periodic tasks.
- **Coupler**: Translates generic commands from the Moral Compass to specific autopilot commands.
- **CVT**: Current Value Table of parameters used as common data set for all EVAA elements
 - **SWIM**: Checks the integrity of data used as inputs

- Monitors

- **GCAS**: Prevents collisions with stationary objects (terrain, features and obstacles)
- **GeoFence**: Prevents intrusion into designated 3-D airspace
- **Forced Landing System**: Detects an inflight fault and lands as soon as possible in the safest location possible.

- Support Functions

- **Map Manager**: Provides map data to the rest of the system.
- **Trajectory Manager**: Prepares a kinematic trajectory prediction.
- **Wind Calcs**: Performs inflight wind estimation based on aircraft track dynamics.

- **Flight Test Module**: Provides an interface to perform flight tests safely.



Findings From EVAA Phase 1A Requirements Development



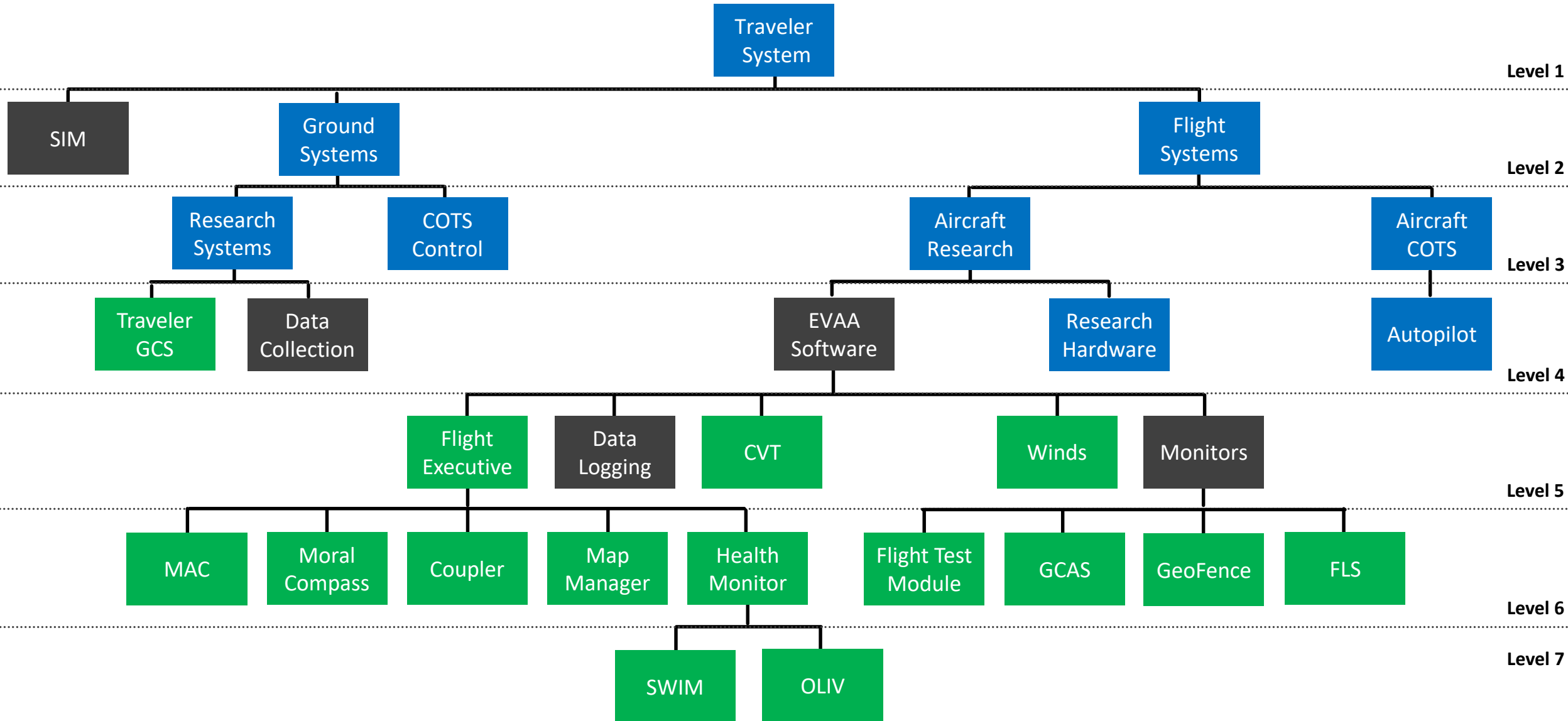
Phase 1A Requirements Development (How Did We Do It)



- Early on the team spent a significant amount of time discussing the system and associated requirements.
 - In the beginning the team had difficulty categorizing requirements. (i.e. appropriate levels within a system decomposition)
 - In retrospect there were multiple systems of requirements being looked at (test enabling systems, EVAA system, etc)
 - A decision was made to proceed with the team's common understanding of the system into design.
 - Part of the research being conducted included developing guidance on development and evolution of requirements for this type of system, however this was not the entirety of the research. Each aspect had to be allotted time and limited to fit within the window.
- As the design progressed requirements were re-visited
 - Flight vehicle platform and associated hardware were previously set
 - EVAA hardware (ODROID) was set
 - Focused on EVAA Software Requirements & Traveler GCS Requirements



Phase 1A Traveler System Decomposition



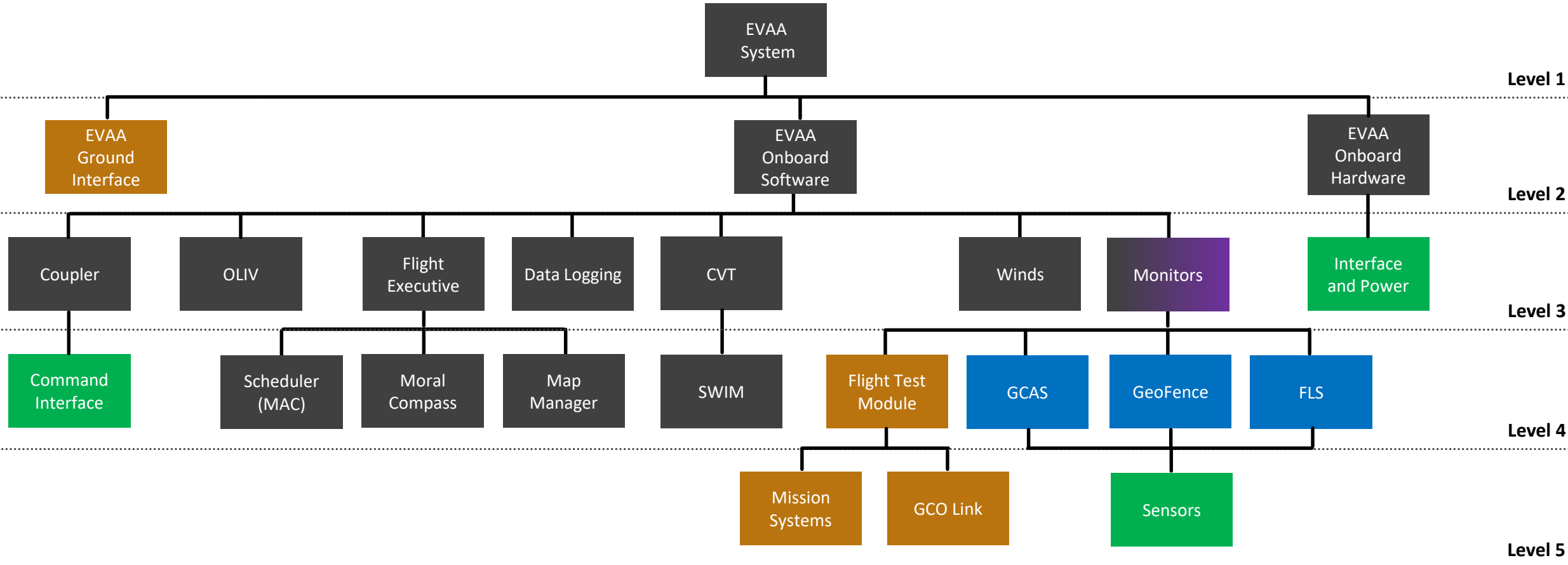


How Could We Do It Better



EVAA System Decomposition

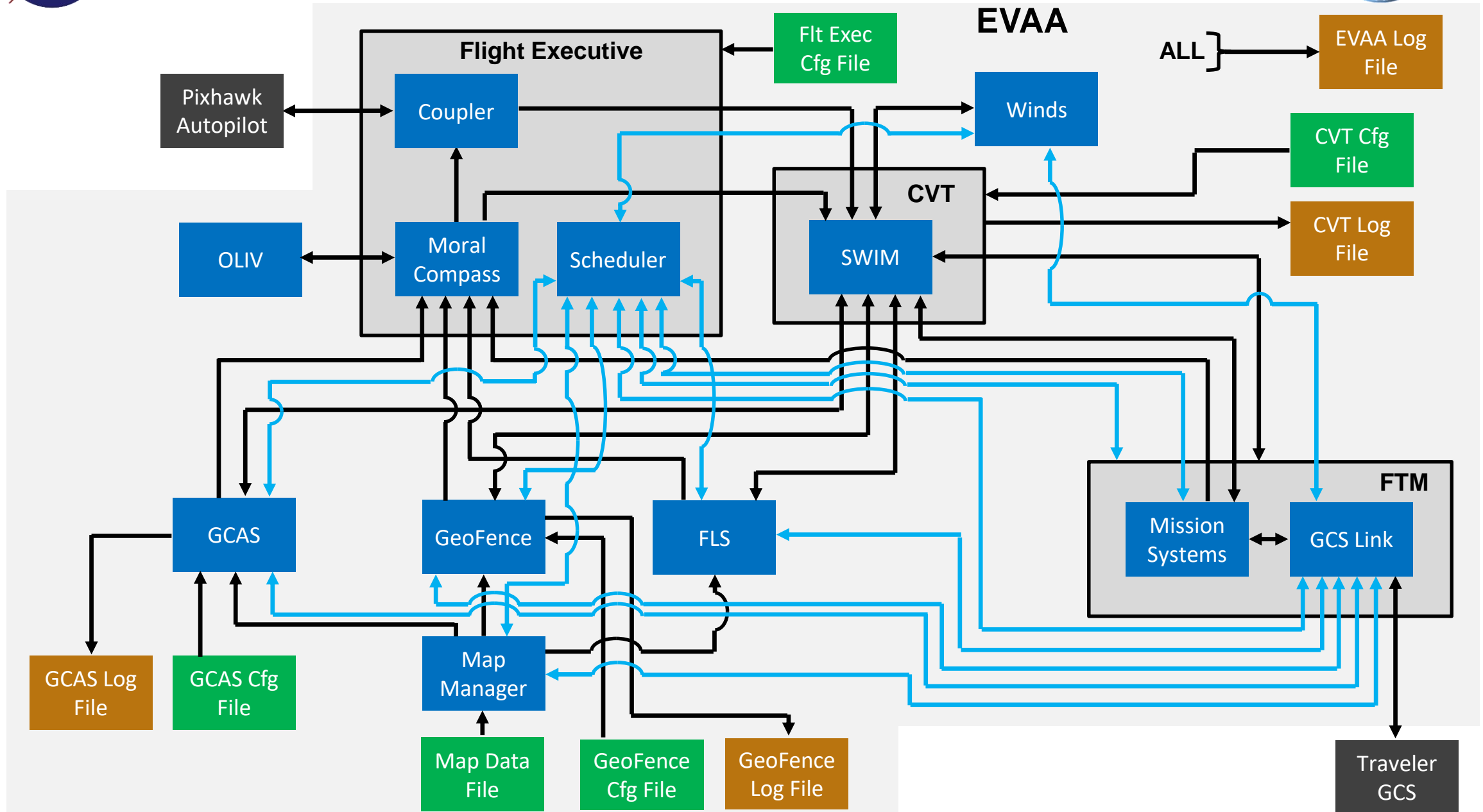
Generic (multiple platform, multiple monitor) System



EVAA Core System
Monitor
Flight Test Specific
Levied on Platform
Levied on Monitors



Data Flow Interfaces





EVAA System Decomposition

Levied Requirements for a Generic (multiple platform, multiple monitor) System



- Monitors
 - Requirements for Monitor developers to interface properly with EVAA
- Levied on Platform
 - Summation of platform specific requirements for EVAA to function properly
 - Hardware interface requirements
 - Mechanical, Power, Communication
 - Sensor requirements
 - Both monitor specific and common state data
 - Command interface requirements
 - Types and formats of commands EVAA must be able to send
 - » Note: For the system to work, all of the autopilot specifics do not need to be provided, what is needed are a few (simple) specific commands and vehicle responses to those commands for avoidance maneuvers.



What Did We Learn – Requirements Development



- Systems of requirements need to marry well (EVAA System, Simulation, Vehicle Platform)
- One common data module / interface is essential for integrating multiple monitors (CVT)
- Configuration files allow for flexibility / expandability
 - CVT JSON file to set list of parameters
 - GCAS / GeoFence config files to set vehicle maneuvers
- Modular architecture with common generic elements allows specific element development / modification with limited regression testing / verification.
 - Strive to be as simple as possible while meeting requirements. Avoid unnecessary complexity in code and design of safety systems
- Monitors should be kept independent (with guidance from requirements on how to interface with the MMRTA architecture)
 - They should not communicate with each other
 - They should each only mitigate one safety risk
 - They should run fast enough to respond faster than the situation can unfold



What Did We Learn – Requirements Development



- Decision logic for monitor fights (Moral Compass) is crucial to ensure safe function and could easily get complex
 - Consequence method (based on non-dimensionalized dollar impact) simplifies this problem
 - Decisions only need to be made at rates which the combined system (MMRTA and vehicle) can respond to
- Flight test enabling requirements need to be carried throughout but kept modular.
 - Need to include flight test specific software (ground and flight) that can be removed after testing complete
- Early consideration of timing constraints is essential.
- Map data storage (indexing and datum) and usage needs to be clearly defined in the requirements provided to monitor developers.
 - Where in a tile does the lat, lon address actually exist?
 - If map lookup area is too small relative to the post resolution, the area can fit between the posts and not capture required elevation data.
 - The horizontal and vertical datum used by the telemetry available from the vehicle should be verified to match the map databases used by the system.



Findings From EVAA Phase 1A Design Development



Phase 1 EVAA Design Effort (How Did We Do It)



- Team decided to use object oriented (C++) language to simplify modularity of design.
 - Typically certified safety critical software would not use object oriented languages due to availability of certified tools for evaluation.
- Software Architecture was established
 - Defined: Base Classes, APIs, Extensible Data Exchange Formats
- Coding style guide and naming conventions were set
- Basic ODROID operating system and packages, and development environment were established
- Major modules were designed first (Flight Executive, CVT, Coupler, and Logger)
- System was broken into software modules then assigned to individual programmers
- Weekly stand up meeting set priorities for modules in work.



Design Philosophy



- Do no harm
 - A Monitor acting alone should not be the cause of an incident
 - With multiple Monitors, tackle the highest consequence first, minimizing harm
- Implementation should accommodate ease of verification
- Solve discrete problems rapidly
 - Simple rules / monitors for specific situations
 - Consequence dictates priority
- Keep the MMRTA system simple and deterministic, push complexity in to common mission manager systems
- Think organization (these types of systems can get complex, some software standards are key)
 - Coding style guide and API guides were essential
 - Naming conventions
- Safety Pilot can always take and keep control during testing



Design Considerations



- Unit testing at the module level prior to integration
- Minimize computational time to enable responsiveness in the system
 - Simplified models used (example: kinematic models)
- Modular architecture with common generic elements
 - Allows specific element development / modification with limited regression testing / verification.
 - Expandability (E in EVAA)
 - Examples: Trajectory prediction and Map Manager are common software elements used by multiple Monitors
- Keep the software simple
 - Each module performs one task or function instead of several (avoid convolution)
- Decision logic for monitor fights (Moral Compass) design/implementation is the critical element that will establish the perceived behavior of the system to the public



What Did We Learn - Design



- Flight test enabling interfaces simplified verification testing / troubleshooting of the system
- The current system can get into avoidance loops
 - EVAA is (as MMRTA should be) safety focused not mission focused
 - The mission manager would / should alleviate this problem, but wasn't the focus of the research.
- Simplicity and modularity in monitor design leads to shorter evaluation / response time which allows for flexibility in the complex function design (i.e. avoids nuisance triggers of the RTA)
 - Each monitor only addresses one risk type
 - Multiple monitors run simultaneously
 - Monitors should not look too far into the future
 - The further out they look the more uncertainty and associated nuisance triggers
 - Dependent upon speed and maneuverability
 - Simple consequence-based Moral Compass – resulted in fast decisions which can accommodate growth to more monitors



What Did We Learn - Design

Certification –Considerations

Safety & Design

- Imperative that MMRTA safety “wrapper” be deterministic (not per DO-178C definition, we mean known repeatable outcome)
 - Let the Mission System employ high-order thinking, complex control ... MMRTA /safety wrapper purpose is to *guarantee* understood behavior / reaction
 - Limitations and assumptions of a MMRTA safety system (or monitor) should be well understood and documented (testers or operators should not be surprised by responses of the MMRTA safety system)
 - Maneuvers, Monitor in control (risk logic), potential monitor force fight loops

Certification Flight Test Design Considerations

- Flight Test Data Requirements must be considered and accounted for in the design
 - Example: Independent from the System Under Test (SUT) position & wind truth sources must be utilized during flight test
- Testing must include off-nominal conditions & should include testing outside operational envelope to evaluate system robustness
 - Winds > mission ops limits > published op envelope; GPS jamming; reduced sensor visibility
 - More off-nominal condition testing should be performed for higher risk operational conditions
- System robustness must be evaluated - Test matrix density, # of data samples
 - Driven by probability and confidence vs. consequence for total risk for a given system



What Did We Learn - Design

EVAA / MMRTA Requirements & Design – Test Ops perspective

- Fully Understand the mission system and aircraft interfaces with the MMRTA system
 - If using a terrain database be sure to understand the impact of the source data upon the MMRTA system design: i.e., is terrain data based on same axis system the aircraft uses?
 - Understand the accuracy of the COTS onboard state-space measurements (Euler angles, Rates, Accelerations, velocities, etc)
 - If the system allows, use of virtual obstacles (GeoFences are by definition virtual) can greatly improve test efficiency and decrease risk
- Determine susceptibility of aircraft trajectory to environmental factors that affect vehicle performance such as winds and density altitude (does system counter winds automatically and stay on desired track/position ... or not ?) and incorporate that consideration into the MMRTA design
- The MMRTA *Monitor* requirements are directly dependent upon the aircraft kinematics for avoidance
 - An aircraft with small turn radius/low speed will likely have different MMRTA monitor/recovery requirements than a vehicle with large turn radius/high speed.



Findings From EVAA Phase 1A Test Approach



Test Approach – Verification (How Did We Do It)



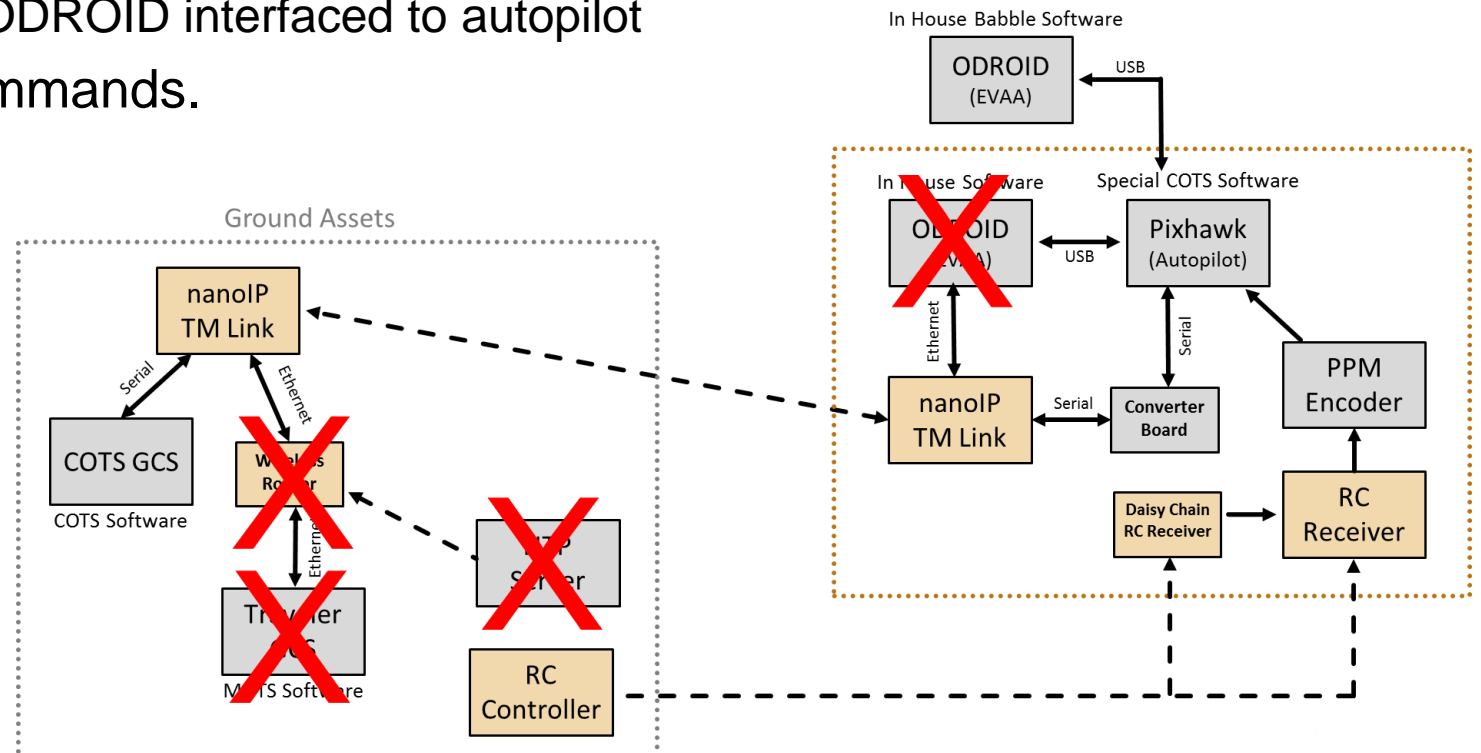
- Unit testing at module level
- Unit testing at integrated level
- Source code inspections performed to supplement testing
- HITL Simulation demonstrations and testing
- Flight hardware ground combined systems testing
 - Babble Testing & Pre-flight
- Flight Testing



Babble / Fuzzer / Load Test



- Test resilience of autopilot (Pixhawk) receiving commands from EVAA hardware (ODROID)
 - Random data, and random commands were sent at max rate to the autopilot
 - Ensure that the autopilot is not overwhelmed with a backlog of commands
 - Ensure that control is not taken from the safety pilot
 - Ensure that the safety pilot can always take control
 - Unique test software on external ODROID interfaced to autopilot
- Testing found multiple problem commands.





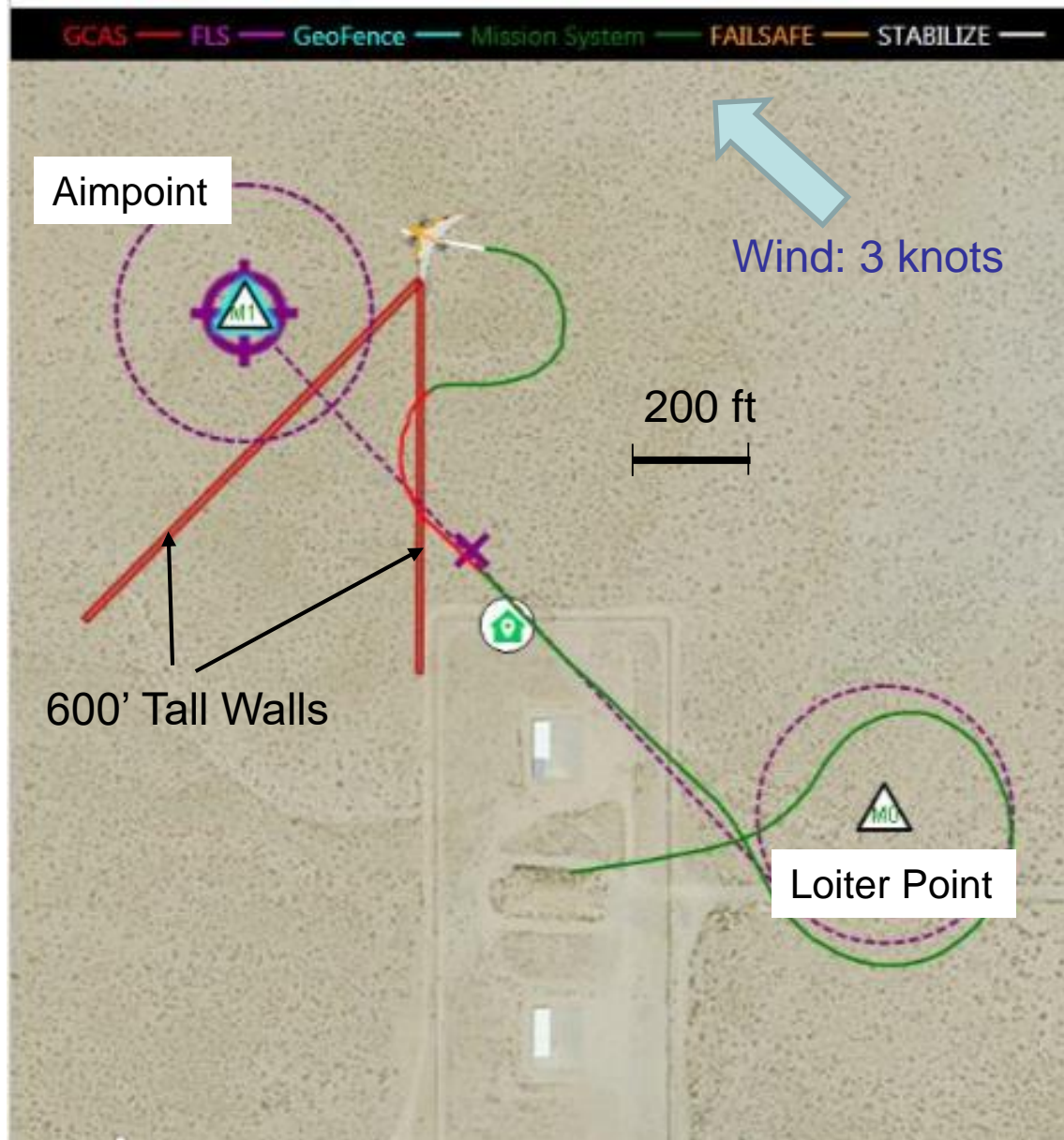
Test Approach – Verification (What Did We Learn)



- Flight test enabling interfaces simplified verification testing & troubleshooting of the system
 - Example: Traveler GCS display allowed for real time assessments of performance in simulation
- Don't assume the "advertised" baseline COTS aircraft performance if adding MMRTA to an existing aircraft – verify areas of need/interest
 - Babble testing to ensure no unforeseen complications in data interface between system under test and COTS autopilot
 - Autopilot control mode switch testing to ensure proper mode function and priority



Test Approach – Flight Test



Mission 24, 18 May 2017 (EVAA #2) Sortie #3

Card #3: GCAS – Avoidance to the Right

- Avoidance: 30° bank, 9 ft/sec climb
- No Horizontal Clearance Buffer
- No wind compensation, penetrated obstacle
- Obstacle penetration: wind?, or aircraft GPS position vs map position?

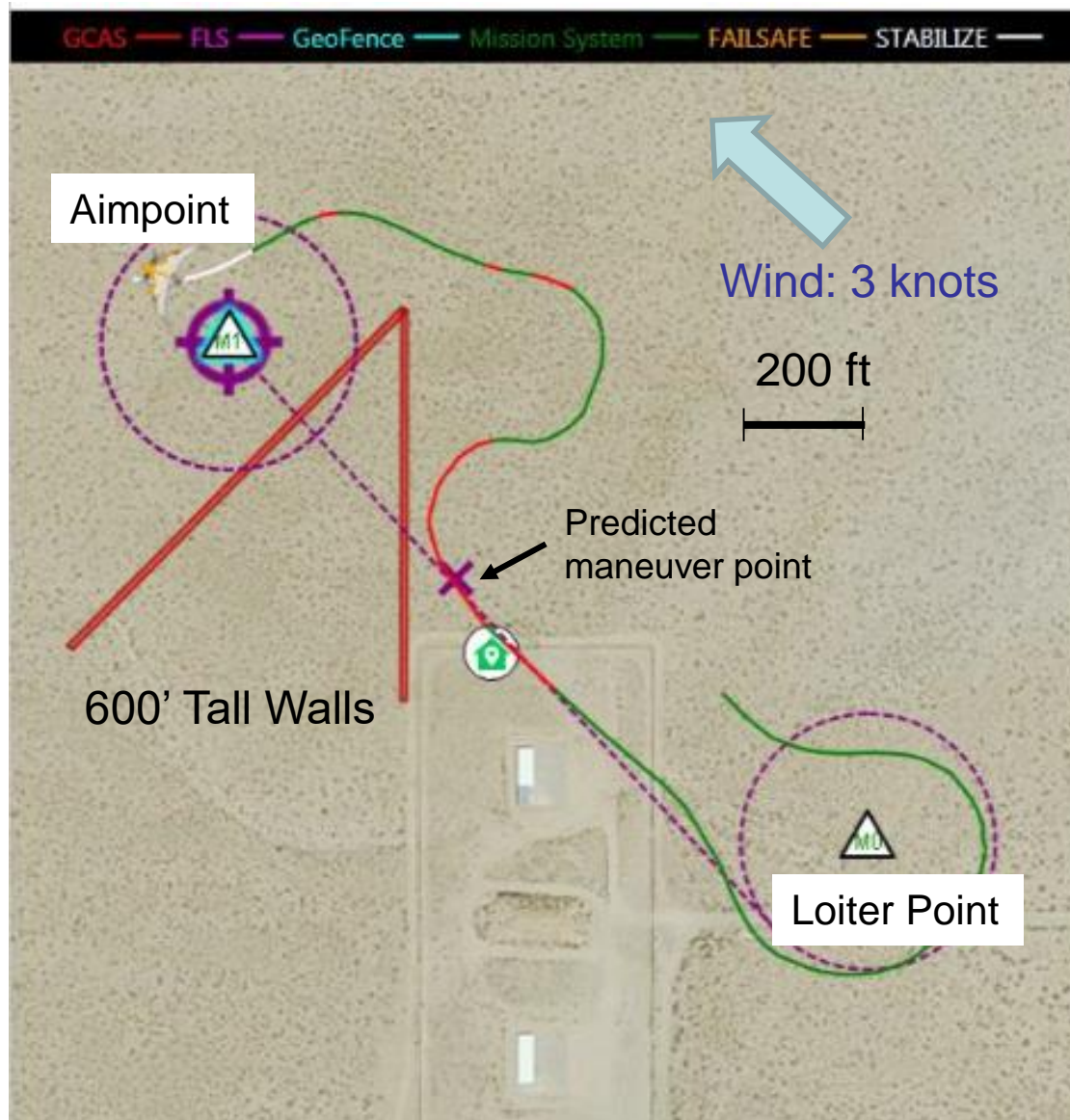
Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)



Mission 24, 18 May 2017 (EVAA #2) Sortie #3

Card #4: GCAS – With 100 ft Horiz Buffer

Avoidance to the Right

- Avoidance: 30° bank, 9 ft/sec climb
- No wind compensation
- Avoided Wall earlier than Card 3 (no buffer)

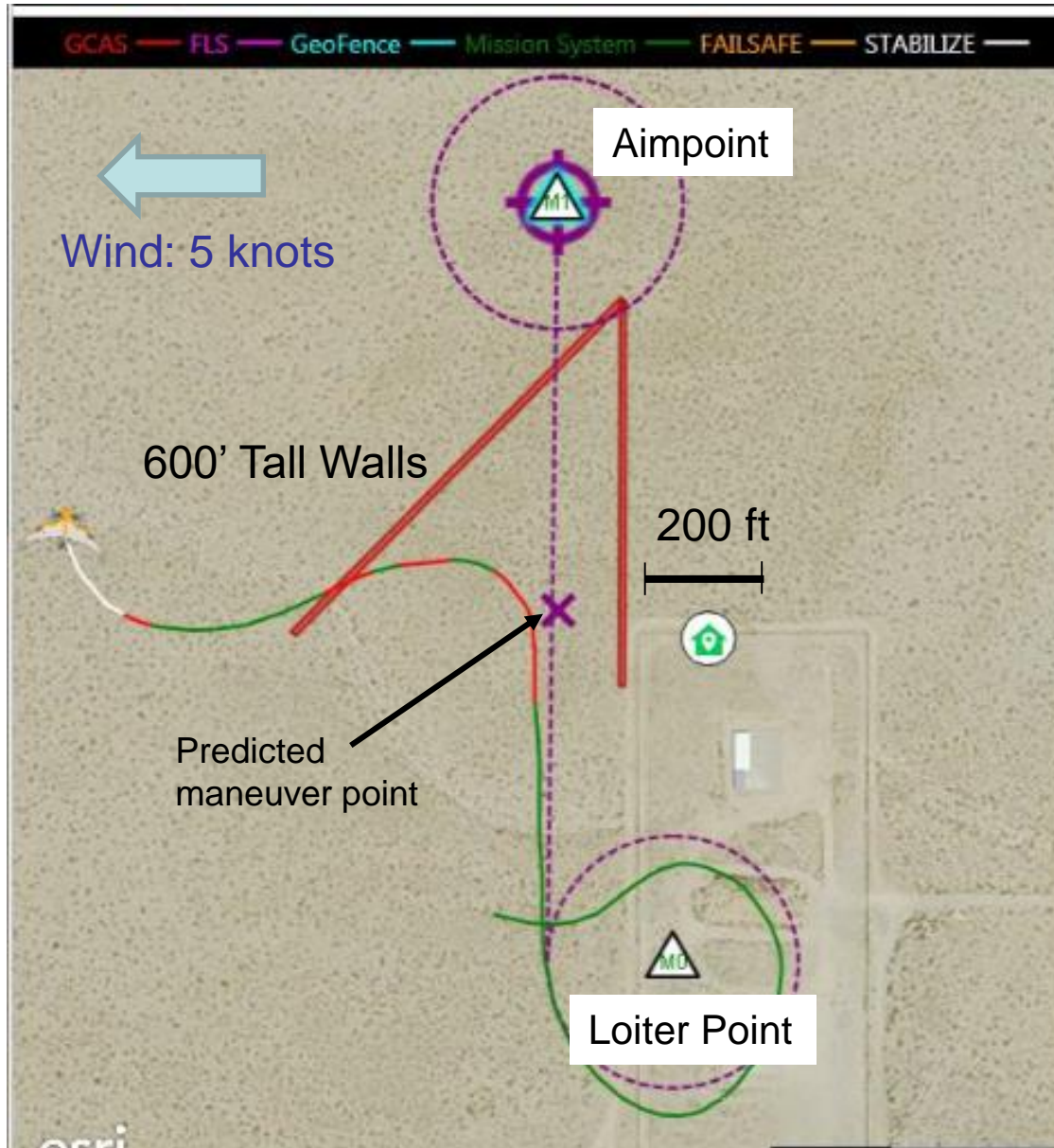
Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)



Mission 24, 18 May 2017 (EVAA #2) Sortie #3

Card #6: GCAS – Pinch Between Walls

- No Buffers
- Avoidance: 30° bank, 9 ft/sec climb
- No wind compensation
- Cause of oblique wall penetration yet to be analyzed

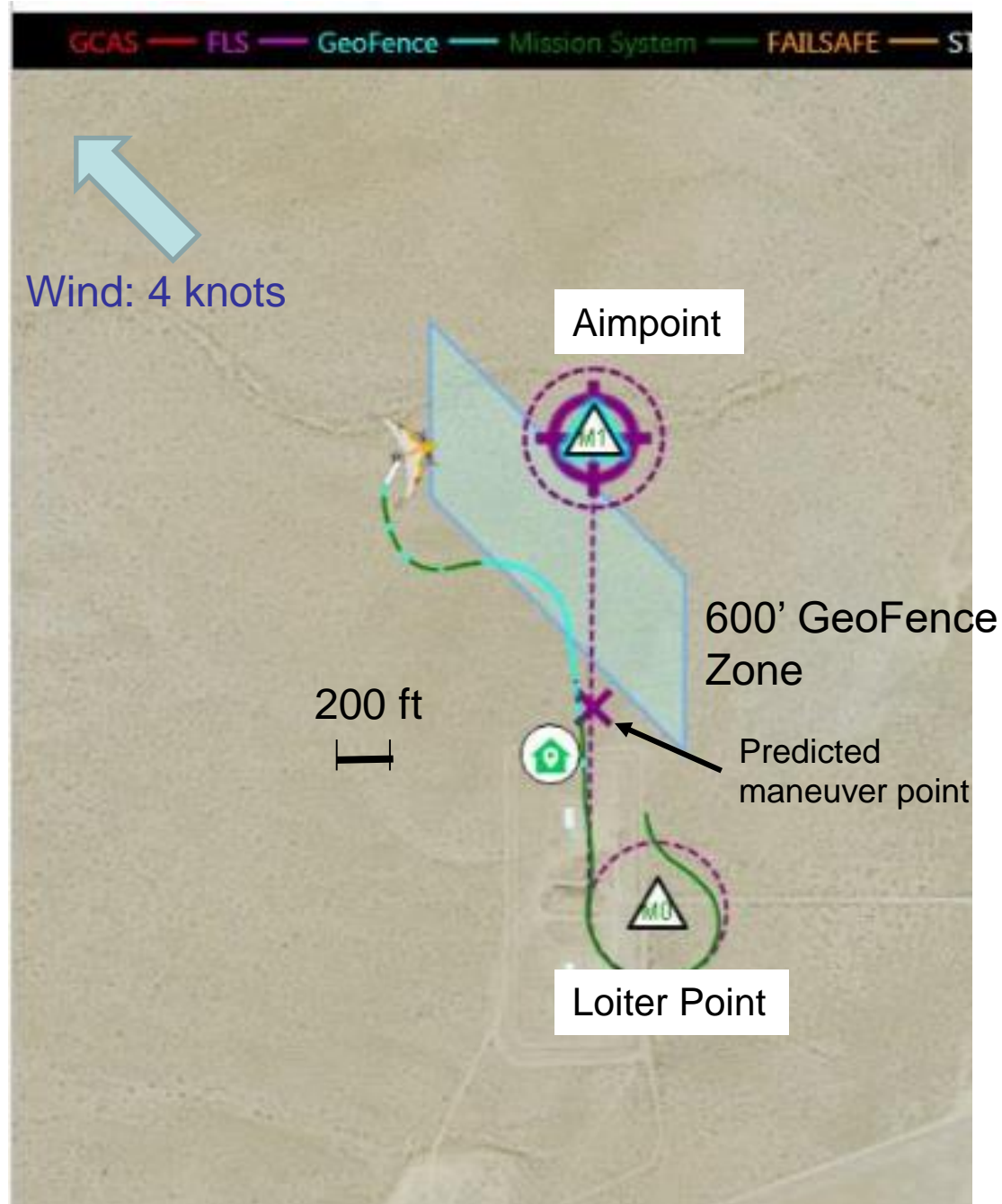
Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)



Mission 24, 18 May 2017 (EVAA #2) Sortie #3

Card #7: GeoFence Avoidance

- No Horizontal Clearance Buffer
- Avoidance: 20° bank, no climb
- No wind compensation
- GeoFence penetration: wind?, or aircraft GPS position vs map position? – yet to be analyzed

Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)

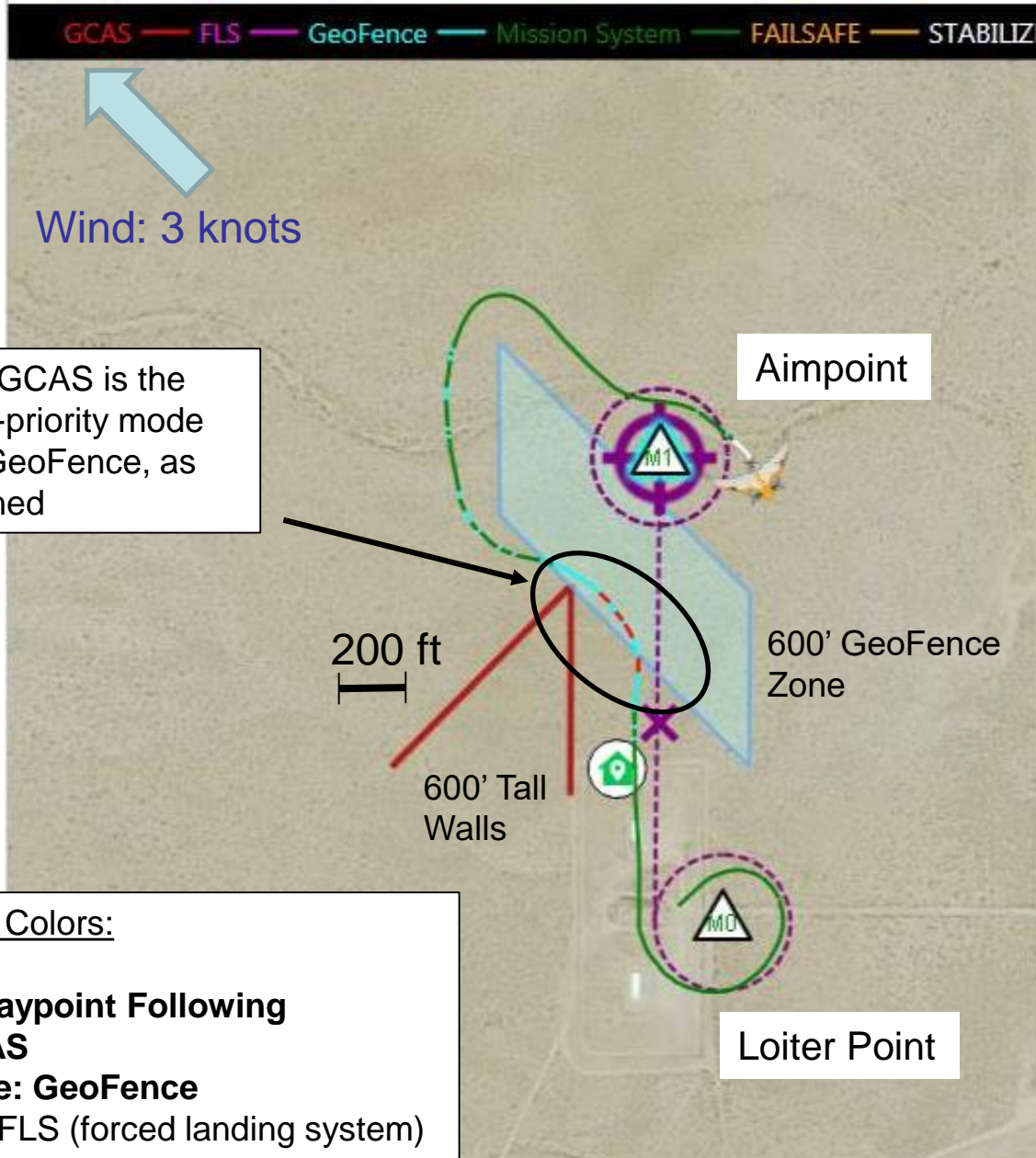


Mission 24, 18 May 2017 (EVAA #2) Sortie #3

Card #8: **1st Multi-Monitor Force Fight (GeoFence 1st)**

GeoFence & GCAS Avoidance

- Per Moral Compass Consequence setting, EVAA should prioritize obstacles instead of GeoFence if both are “active”
- 50' Horizontal Clearance Buffer on GeoFence, 0' on Obstacle
- GeoFence Avoidance: 20° bank, no climb
- GCAS Avoidance: 30° bank, 9 ft/sec climb
- No wind compensation
- Maneuver yet to be analyzed.



Trajectory Colors:

Green: Waypoint Following
Red: GCAS
Light Blue: GeoFence
Magenta: FLS (forced landing system)



Mission 24, 18 May 2017 (EVAA #2) Sortie #4

Card #10: Multi-Monitor Force Fight (Obstacle 1st)

GeoFence & GCAS Avoidance

- 0' Horizontal Clearance Buffer on GeoFence, 50' on Obstacle
- GeoFence Avoidance: 20° bank, no climb
- GCAS Avoidance: 30° bank, 9 ft/sec climb
- No wind compensation
- Obstacle penetration cause: yet to be analyzed

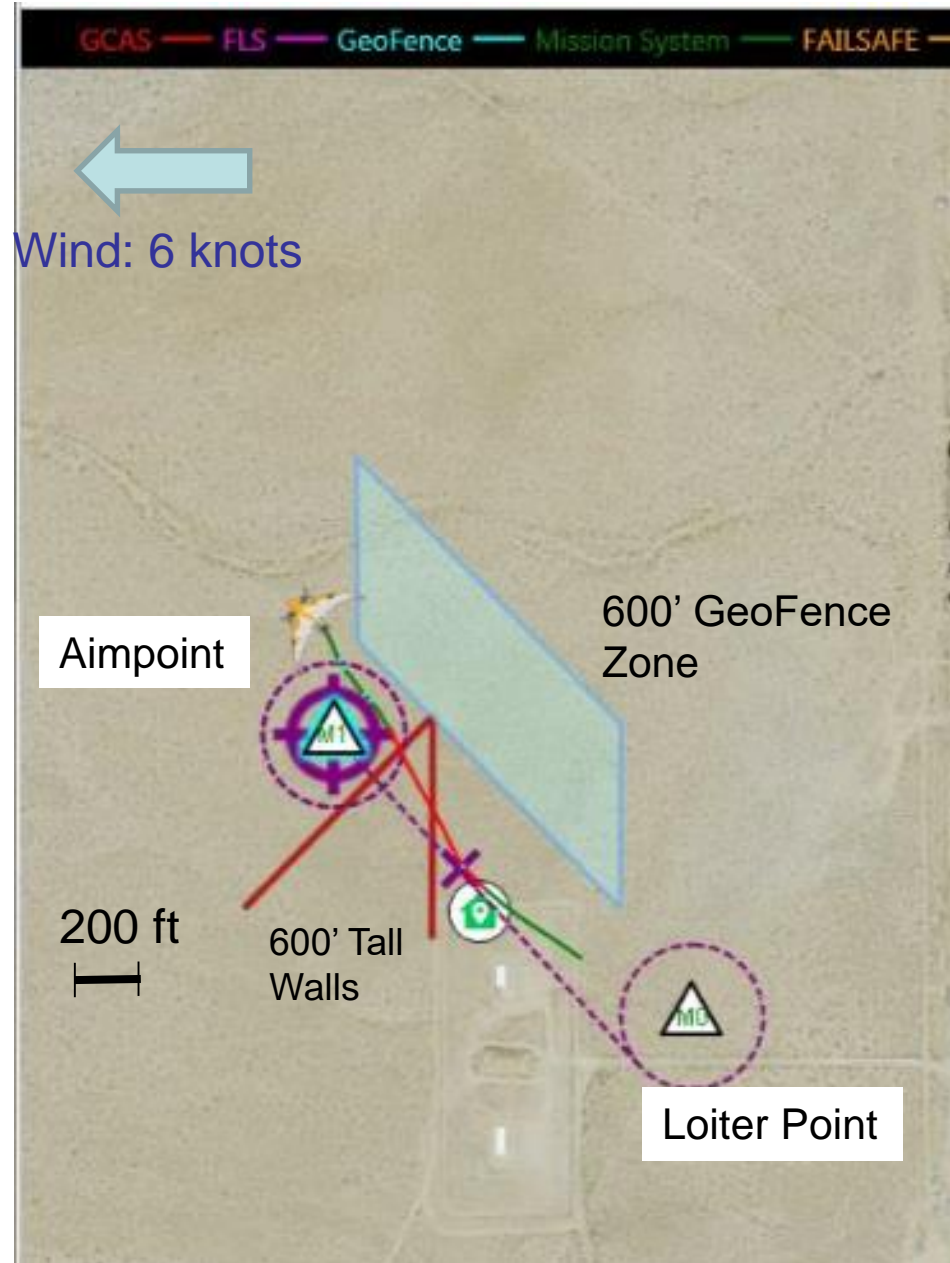
Trajectory Colors:

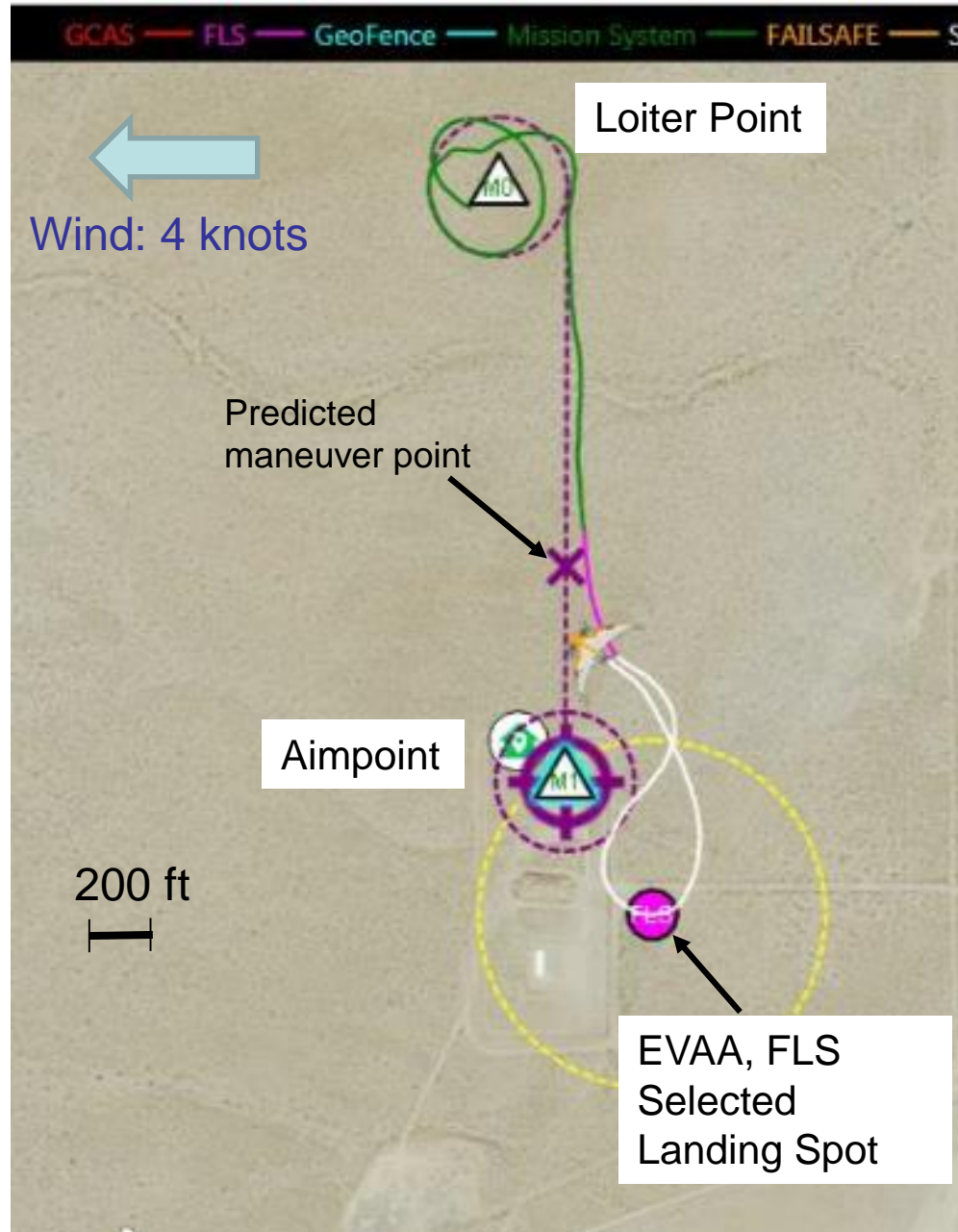
Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)





Mission 24, 18 May 2017 (EVAA #2) Sortie #4

Card #12: FLS - Immediate Land Mode

- Looks 10 sec out, lands on lowest cost spot within 200 ft radius along aircraft's true heading at time of initiation
- FLS determined landing point was not a good location (sagebrush) (white path is pilot taking control)
- No wind compensation

Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)

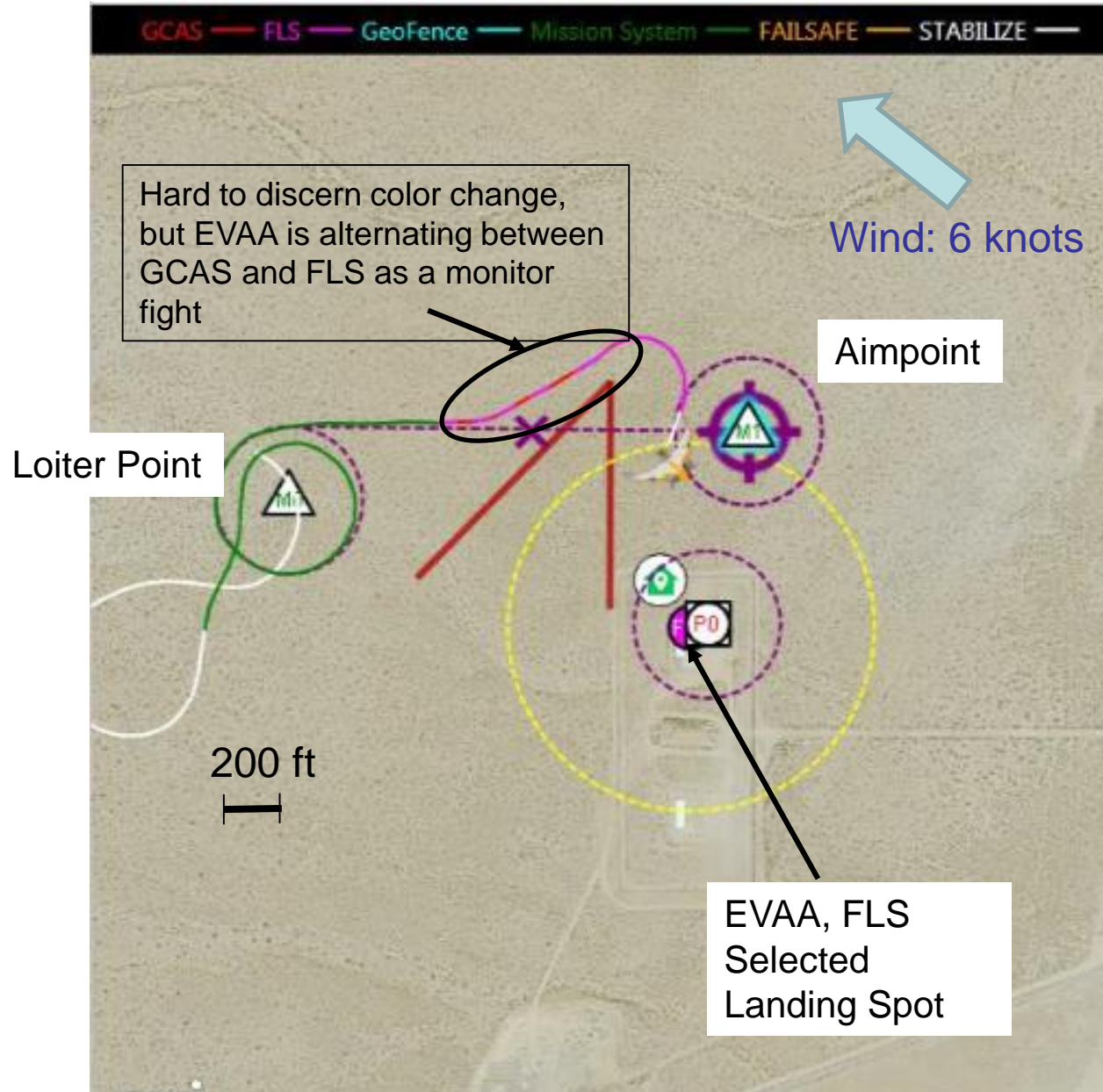
White: pilot control



Mission 24, 18 May 2017 (EVAA #2) Sortie #5

Card #14: Multi-Monitor: FLS-land here Mode vs GCAS

- Per Moral Compass Consequence Setting, Obstacles should (and were) avoided during FLS
- FLS-land here mode: landing spot selected by Ground Control operator
- No wind compensation
- 50' Obstacle Horizontal Clearance Buffers



Trajectory Colors:

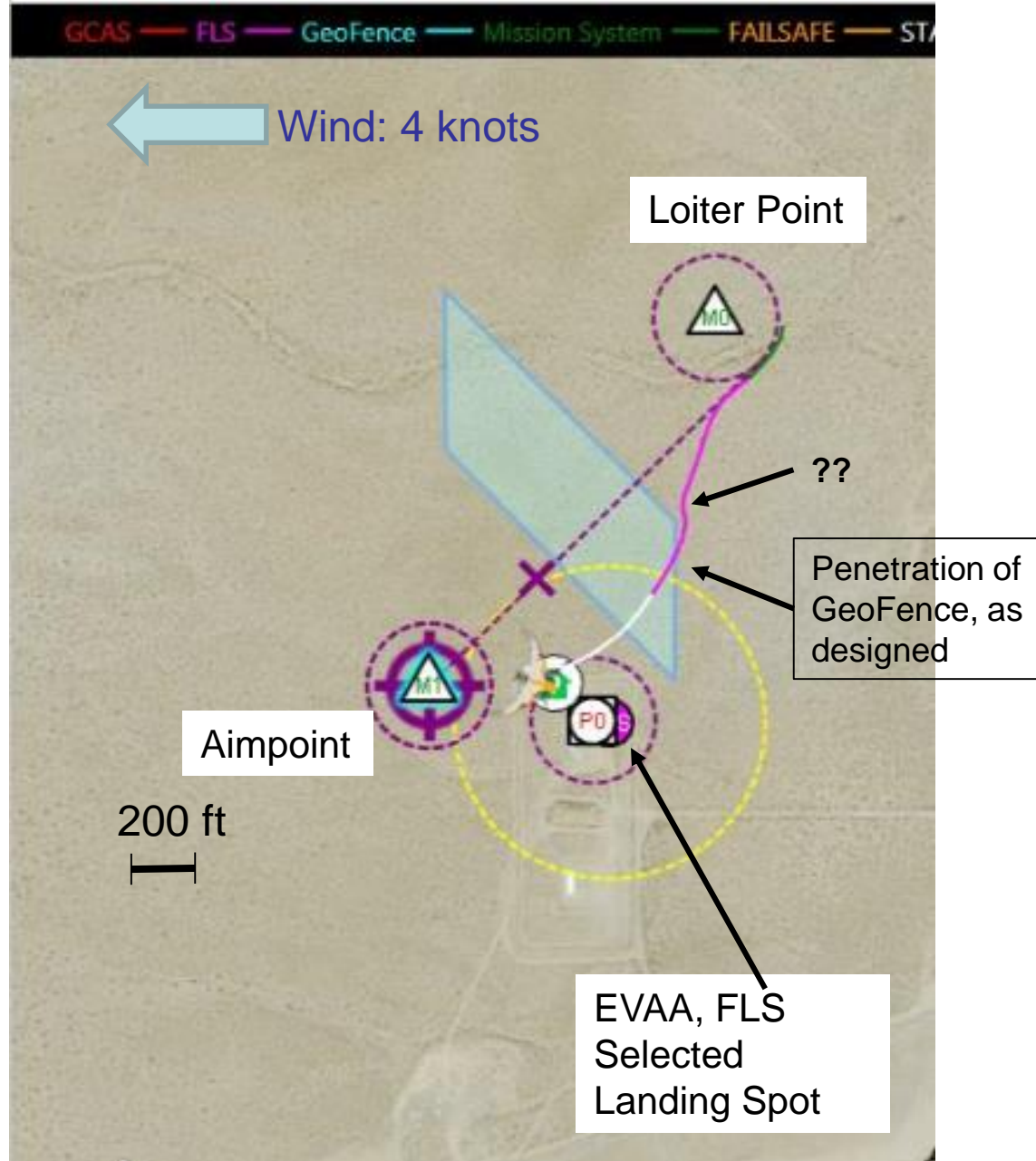
Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)

White: pilot control



Mission 24, 18 May 2017 (EVAA #2) Sortie #5

Card #15: Multi-Monitor: **FLS-land here Mode vs GeoFence**

- Per Moral Compass Consequence setting, Aircraft should (and did) penetrate GeoFence during FLS – note trajectory colored remained Magenta (FLS = highest priority)
- FLS-land here mode: landing spot selected by Ground Control operator
- No wind compensation
- No GeoFence Buffer

Trajectory Colors:

Green: Waypoint Following

Red: GCAS

Light Blue: GeoFence

Magenta: FLS (forced landing system)



Mission 24, 18 May 2017 (EVAA #2)
Sortie #5

Card #16: Triple Multi-Monitor: FLS-
land here Mode vs GeoFence vs
GCAS

- Unfortunately, did not initiate a screen capture



What Did We Learn – Flight Test



Test Planning

- Don't assume the “advertised” baseline COTS aircraft performance if adding MMRTA to an existing aircraft
 - verify areas of need/interest
 - Compare aircraft envelope against Mission ops demand. Example: If operating in Denver, consider Density-Altitude effects (see supporting briefing slides)
 - Wind effects upon departure susceptibility, avoidance maneuvers, cruise performance
 - Understand the accuracy of your COTS onboard state-space measurements (Euler angles, Rates, Accelerations, velocities, etc)
- Don't always assume aircraft performance is same across fleet
 - Vtrue & roll response differences between aircraft - Affected avoidance maneuvers and inflight wind calculations - resulted in need for tail number-specific configuration file in software
- Plan test profiles that will stimulate the software /avoidances in multiple ways / varying setups
 - Otherwise risk missing unanticipated system responses
 - NASA-Traveler has ~100 test profiles for 3 Monitors (GCAS, GeoFence, & Forced landing System)
- A Simulator /model can be very helpful in MMRTA software development, test matrix planning, and test mission prep, but NOT a valid substitute for MMRTA flight test
 - Req'd fidelity of aircraft kinematics in sim/model depends upon the missions ops
 - Example: if not in constrained flight environment (not closely space multi-obstacles) then High-fidelity of aircraft kinematics is likely NOT necessary
 - But be careful to ensure data inputs to MMRTA within the sim/model match those provided from the actual flight autopilot/system hardware/software (axis systems, units, sign, etc)



What Did We Learn – Flight Test

Test Planning – cont'd

- Truth Source Data – KNOW the limitations of the aircraft parameters / data
 - Position (lat / long / altitude) – Must understand your onboard-system's accuracy, source (GPS aided?), drift, etc
 - This will drive whether independent / ground-based position measurement system is also needed
 - Understand your earth-position axis system (i.e., is it WGS -84 or another system ??)
 - Baro / pitot air data (altitude, speed, etc) requirements for MMRTA system and data analysis depend upon aircraft operations: example: pitot-tube derived airspeed may not be required if aircraft flies at constant airspeed (compute V_{true} from measured / fixed ground speed and atmosphere conditions)
 - Consider calculating winds aloft – in majority of cases they are necessary or very helpful to MMRTA.
 - Essentially a MMRTA system design tradeoff of avoidance clearance vs nuisance avoidance
 - Need driven by aircraft avoidance kinematics coupled with mission operations (i.e., are winds needed if flying between bldgs.? Ground trajectory and distance of closest avoidance is function of [vehicle speed / wind speed])
- Minimize the baseline/COTS System reliance upon the System Under Test (SUT) for test safety and impact upon the COTS system performance
 - Otherwise extremely difficult / impossible to isolate the SUT / MMRTA effects from the baseline
- Aircraft characteristics & operations will affect MMRTA data needs and test-analysis data needs
 - For instance: Does aircraft fly constant airspeed Or constant ground speed ? Effect of consumables upon performance (e.g., fuel weight effect on turn performance) ?



What Did We Learn – Flight Test



Test Execution

- In most instances sUAS flight test efficiency possibly more productive and almost as efficient as modeling/simulation
 - Accurate modeling of most sUAS flight dynamics could be more costly than flight test
 - Non-linear, or unpredictable influences are very difficult to model (wind eddys around buildings)
 - Generally, sUAS flights are low cost, quick-turnaround, efficient, safe, etc
 - Use virtual obstacles (GeoFences are by definition, virtual) greatly improved test efficiency and provides enormous test flexibility
 - Did not have to travel to various ranges to utilize limited size/shaped terrain or obstacles
- Utilizing a safety pilot (SP) greatly reduces risk during testing
 - Ensure the Pilot-take-control is independent of the SUT (Safety not provided by SUT during evaluation)
 - Verification of Safety Pilot taking and keeping control is crucial (babble / mode switch testing)
- Plan for, & Gather data useful for postflight analysis that provides trend data over long periods of time or many flights that may /may not be useful for individual flights
 - Examples: Obtain independent (i.e., not from SUT) 3D position accuracy (not only during test profiles, but also takeoff AND landing to assess drift error over time), number and type of avoidance maneuvers during sorties, time history of winds aloft correlated with aircraft time, distance of closest approach during avoidance maneuvers



What Did We Learn – Flight Test



- Flight test enabling interfaces simplified verification testing / troubleshooting of the system
 - Implemented ability to turn monitors ON/OFF/Standby, thus providing improved test efficiency and better control of test buildup
 - Improved ability to use virtual obstacles & GeoFences that could be “de-activated” when not needed in order to prevent interference
 - Standby allows data capture without vehicle response



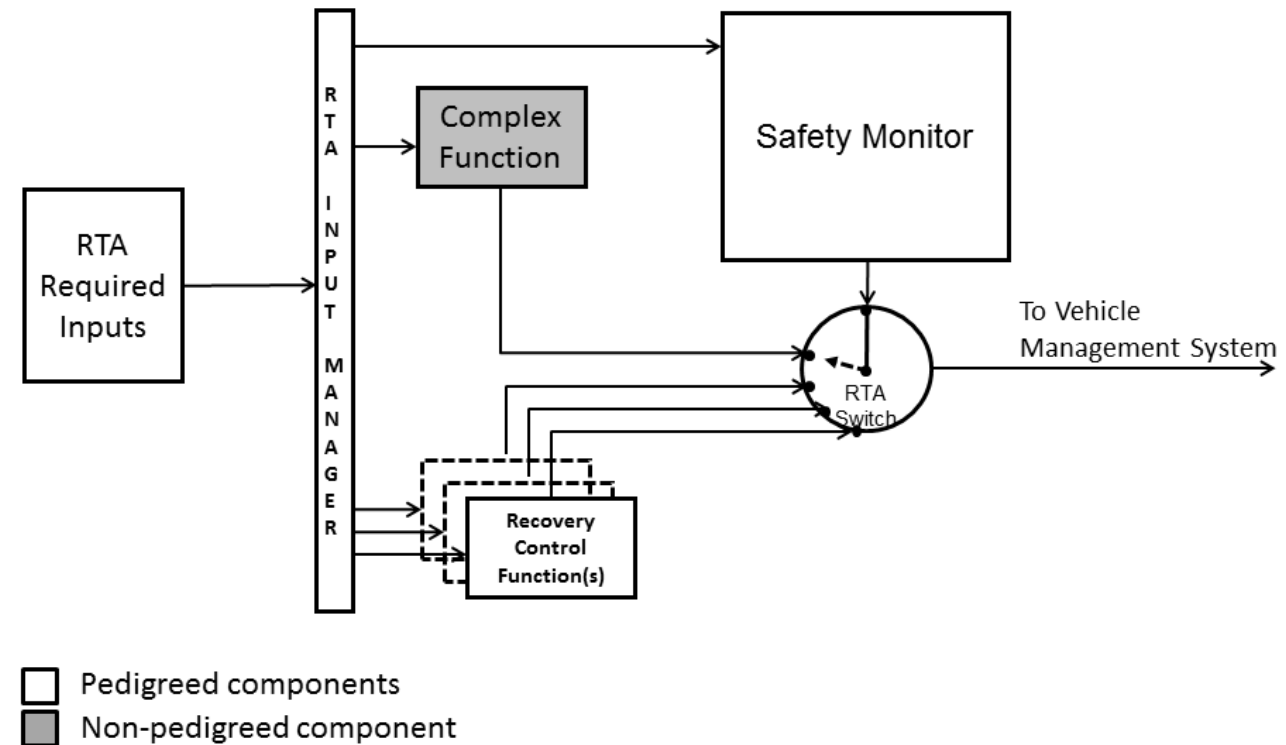
Standard Practice Review

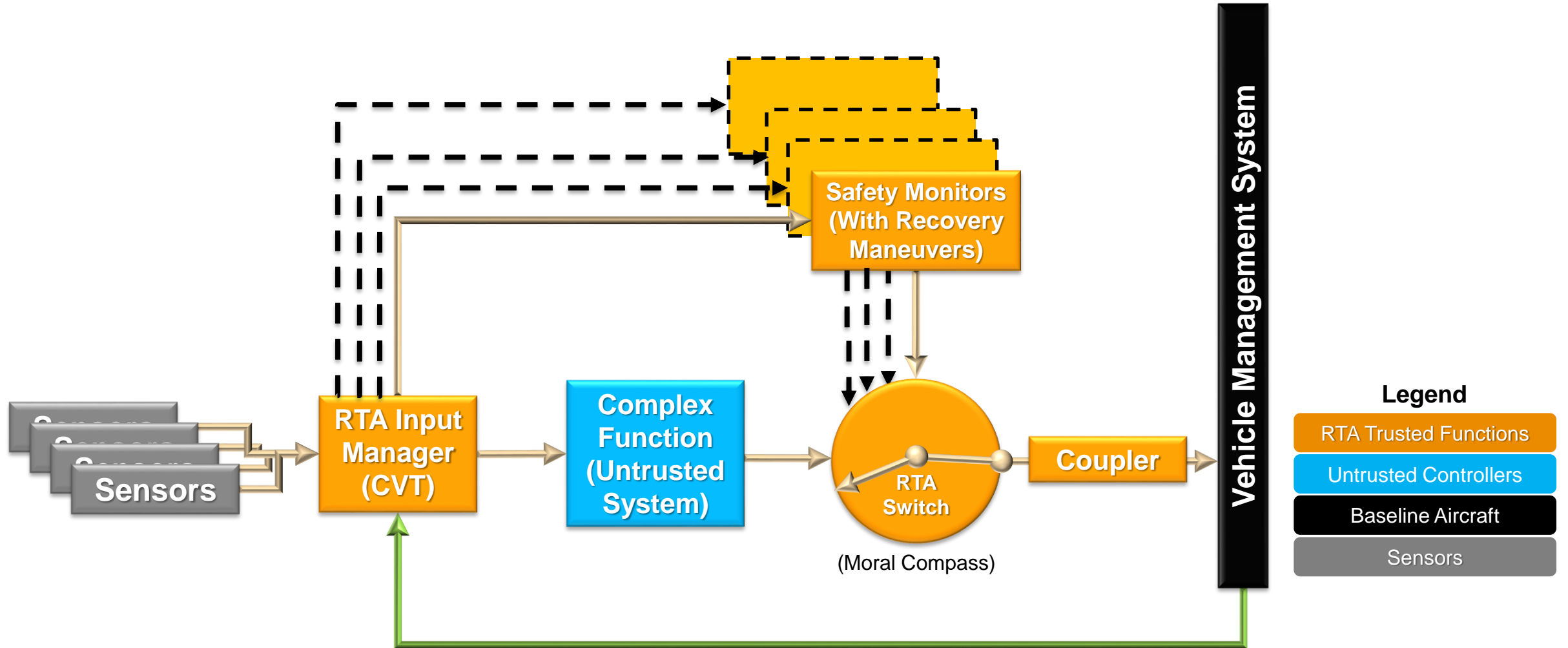


Review of ASTM Standard Practice



- Current standard is overly prescriptive to one design solution with only one possibly complex monitor.
 - Several current requirements drive one monitor architecture specifically.
Example 5.6.1.1 “The RTA Switch shall receive a command from *the* Safety Monitor to use either the CF or one of the Recovery Control Functions.”
 - Current EVAA architecture contains multiple monitors, each send a recovery command and consequence. (each monitor utilizes the predicted trajectory of its recovery commands to trigger) The moral compass implements the highest consequence recovery.
- One monitor solution could result in the monitor being a complex function.





*Note: Recovery maneuvers are contained within monitors, but may be updated via configuration file for modularity / adaptability across platforms.



Review of ASTM Standard Practice



- General recommendations:
 - Eliminate requirements that prescribe one design solution
 - Include design guidance section to fill gap between general requirements and recommended practices
 - Include certification verification guidance section for the same reasons
 - See lessons learned from testing / evaluation
 - Include checks on recovery functions to ensure appropriate for the condition
 - The document reads as though only UAS manufacturers would build these types of systems. Third party entities are not accounted for in the guidance and should be. (Applicants may not be UAS manufacturers)
 - More detail should be provided on monitor performance vs complex function performance
 - Avoid Nuisance RTA triggers i.e. Monitors should not look too far into the future / react so far out they limit the complex function unnecessarily



Review of ASTM Standard Practice



- 1.2 – “enabling highly automated UAS operations” ... to include autonomous?
- 1.3 – “Certification challenges under DO-178C include generating required artifacts, such as requirements, elimination of unintended functionality, traceability/coverage, and test cases required for verification.” ... the issue isn’t in requirements definition, but instead adequate test case definition.
- 1.5.5 – “An RTA design with multiple RCFs should consider the aircraft state, relative outcomes, and differences in RTA recovery times in prioritizing the recovery actions in the Safety Monitor.” ... include aircraft kinematics (might be what was meant by relative outcomes)
- 3.1 Definitions - “Safety Monitor – Continually monitors aircraft state including performance and functioning of the complex function to determine if the aircraft is or is predicted to exceed Pre-defined Limits. As necessary it will control the safety switch” ... think you mean RTA Switch
- 5.1.3.1.2 – “The UAS manufacturer shall consider the impact to risk from the an incorrect action commanded by the Safety Monitor, RTA Switch, and/or the Recovery Control Function(s).” ... consider is unverifiable / open ended
- 5.1.5.1.1 – “Run-time assurance architecture” ... assuming you mean architecture description document



Review of ASTM Standard Practice



- 5.1.5.2.2 – “Simulation and/or flight test results from anticipated environmental conditions and missions where the Recovery Control Function(s) are not expected to activate.” and 5.1.5.2.3 – “Simulation and/or flight test results of stressing cases that cause the activation of all Recovery Control Function(s) relative to the Concept of Operations.”
 - There are some significant limitations to simulation of sUAS, more is needed to substantiate function.
- 5.1.5.4.2 - “Methods acceptable to the CAA **may** be used to identify the artifacts required for complex airborne electronic hardware.” ... Confusing as CAA acceptable methods would by default have to be complied with
- 5.2.3.1.1 – “The UAS manufacturer should take into account the vehicle dynamics, environmental dynamics, and Safety Monitor thresholds in determining the frequency, accuracy, and precision requirements of each data element.” ... depending on the data element this might need to be a shall (whether or not it feeds safety critical functions)
- 5.3.2.4 – “The RTA Input Manager should calculate and report estimated uncertainty of the measured state of the UAS resulting from deviations in precision, accuracy, and latency of the state variables.” ... overly prescriptive, uncertainty must be accounted for, but there are other ways to account for uncertainty than in the RTA Input Manager



Review of ASTM Standard Practice



- 5.3.3.2 – “The UAS manufacturer should take into account the vehicle dynamics, environmental dynamics, and Safety Monitor thresholds in determining the precision, accuracy and latency requirements for each state variable.” ... depending on the state variable this might need to be a shall (whether or not it feeds safety critical functions)
- 5.4 - Add an item that requires ‘sufficient isolation’ between the complex function and the safety monitors. The intent is to prevent the complex function from corrupting safety monitors processing or starving safety monitors of processing resources.
- 5.5.2.1 – “Each RCF shall be designed to provide a set of commands to account for contingencies that can occur in UAS operations.” ... overly prescriptive to a particular design architecture (RCFs external to monitors), also requires each to handle all cases (specific contingencies)
- 5.7.2.3 – “The Safety Monitor shall determine the RCF Trigger Thresholds based on the RTA Recovery Time for each RCF.” ... Overly prescriptive to a particular design architecture. Monitor with recovery functions included accounts for maneuver times to trigger appropriately
- 5.7.2.3.1 – “The RTA Recovery Time shall be calculated for each RCF based on UAS state and environment.” ... Overly prescriptive to a particular design architecture. These don’t have to be calculated real time it can be a look up table or worst case value



Review of ASTM Standard Practice



- 5.7.2.8.1 – “Each RTA Switch transition shall be timestamped and included in the recorded data to enable analysis of RCF reliance and nuisance RCF activations.” ... need to include vehicle management system data as well.



Considerations for Autonomous Systems Certification Framework / Approach



sUAS Functional Requirements



- Intersection of sUAS Functions (Aviate, Navigate, Communicate) with Autonomy is dependent upon the sUAS ConOps
 - How/Where/When the sUAS is to be operated (i.e., its ConOps) drives risks. Mitigation of these risks is accomplished in the 3 functional areas:
 - Aviate
 - Avoidance: Midair, terrain/obstacles, severe weather, & people/animals
 - Degraded Control addressed at later datebut contemplating capturing degraded control reqmts in categories that are dependent upon the Conops risk zones
 - Basic Airworthiness (controlled flight & recovery) are independent of Autonomy safeguards
 - Communicate
 - Coordination with Airspace controlling agency
 - Coordination with operator's mission operations
 - Navigate
 - The “exposure” of the vehicle to the public due to Conops should also influence the rigor and processes for hardware & software development
 - Hardware (e.g., DO-254), Software (DO-178B), and QA processes
 - Consider certification test matrix density as it relates to the overall system risk (think what probability with what confidence versus what consequence)



ConOps – Influences Upon sUAS Reqmts



- ConOps – How, Where, & When the sUAS is utilized has major influence upon the sUAS Functional requirements. **ConOps can be Categorized into 2 dimensions:**
 - **Mission Execution** – what is the risk posed to people, obstacles, other vehicles as a result of the Mission ConOps?
 - How the risks are mitigated drives the cert requirements and accomplished through the Functional Categories (Aviate, Navigate, Communicate)
 - **Environment** – all weather? Day/night? Winds? also drives the certification requirements

Overlay the Impact of ConOps on Functional Needs results in System Reqmts

Functions

Midair Collision Avoidance

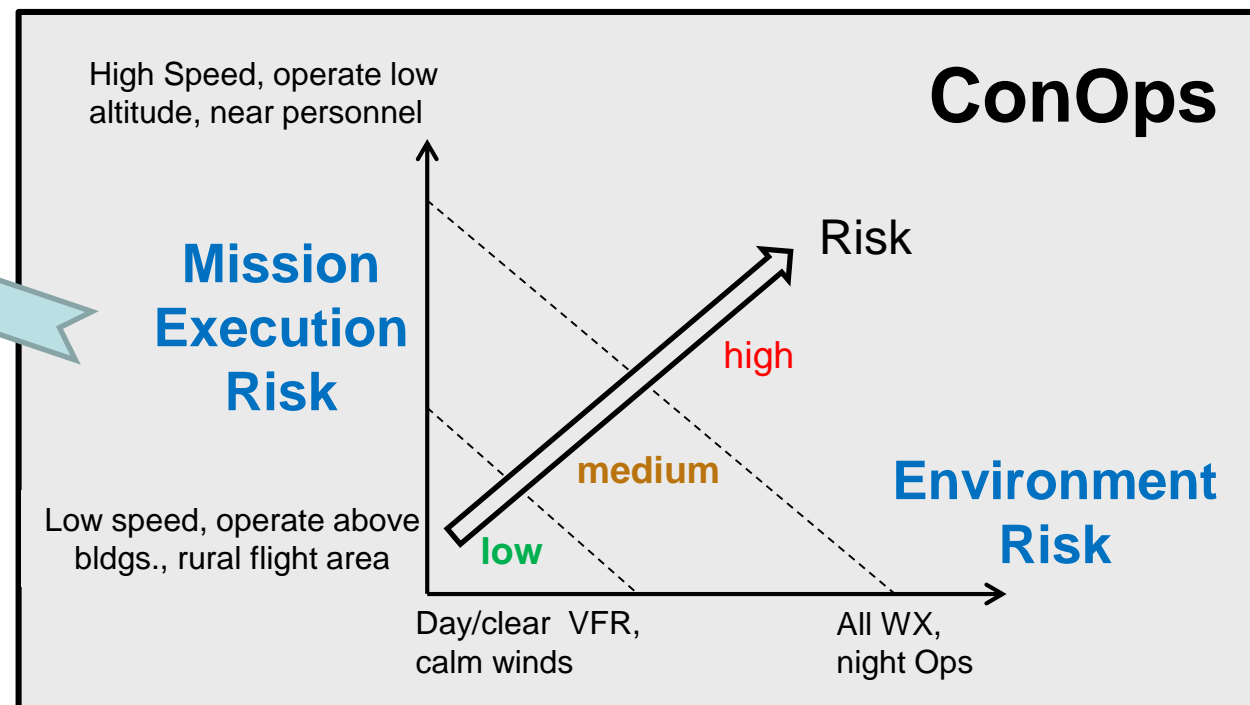
Terrain/Obstacle Avoidance

Severe Weather Avoidance

People/Animal Avoidance

System Reqmts

ConOps





sUAS Cert – One Approach: a Point/or Probability-Based Matrix



- Specify minimum # of points (*or probability*) that must be achieved in various Cert *Categories*
 - The Categories pertain to the sUAS Functionality: **Aviate, Navigate, & Communicate**
..... **And**
 - The Categories are also dependent How / Where the sUAS is to be utilized (i.e., **Conops: Mission Execution & Environment**)
- The Categories form a 3-D cross matrix of *Functionality (1-D)* vs *ConOps (2-D)*
 - ASTM / FAA must specify how the “points” or the necessary probabilities (with associated Confidence values) are achieved within each “cell” of Cert Matrix
- This method provides freedom to the Cert Applicant on how they want to satisfy the Cert Requirements
 - Make sure you protect against the requesting OEM “gaming” the MMRTA intent via narrow, non-robust implementation in 1 or more areas.
 - Part of new sUAS /FAR 23 cert system should consider weaving robustness into the reqmts

3-D
Matrix



Autonomy Cert Matrix- Example

Conops: Mission Execution



Functional Requirements

Functional Reqmts - Autonomy Dependent

			Mission Execution	
			Actual	GOAL
Aviate	Flight Conduct			
		Midair Collision Avoidance		
		Ground/Obstacle Avoidance		
		Severe Weather Avoidance		
		Personnel / Animal Harm Avoidance		
	Degraded Control			
		Thrust		
		Lift		
		Roll/Pitch		
		Sensing Processing		
Communicate				
	Coord with airspace control			
	Coord with mission operations			
Navigate				
	Mission Execution			

Conops, Environment:
(day/night, rain, fog. etc): Overlay Weather considerations with Functional Reqmts & Mission Execution Conops

Mission Execution: Within each "Actual"-Cell , the cert applicant calculates the # of points OR a probability value achieved

Mission Execution: Within each "Goal"-Cell , prescribe a minimum # of points OR a probability value that must be achieved (the Cert-Goal value to be achieved is a function of the ConOps).

		Mission Execution - Actual	Mission Execution - GOAL	Environment - Actual	Environment -GOAL
Flight Conduct					
	Midair Collision Avoidance				
	Ground/Obstacle Avoidance				
	Severe Weather Avoidance				
Degraded Control	Personnel / Animal Harm Avoidance				
	Thrust				
	Lift				
	Roll/Pitch				
	Sensing				
	Processing				
Coord with airspace control					
Coord with mission operations					
Mission Execution					

Matrix Example of 1 aspect of ConOps: Mission Execution - Persons / Animals / Cars Avoidance

Goal: <0.02%

Achieved: <0.0125%

Calculate probability of striking person/animal/vehicle

- Exposure
- Kinetic Energy (speed,weight)
- Avoidance procedures
- Aircraft lighting/warning to people
- EVAA performance

	Exposure Per Op or Fly Hr?	Mission Execution Risk Probability				Achieved	Goal
		Kinetic Energy	Procedural	Deconfliction/Behavioral	Avoidance/Safety		
Midair							
Ground/Property Impact							
People/cars/animals	1.000%	50.000%	50.000%	50.000%	10.000%	0.01250%	0.020%
Navigate							
Communicate							

probability aircraft will strike person (thius example calculate using exposure area ratios).
(Radius of tent area)**2 /
(radius of aircraft flight path
with little/no reaction time)**2:
(10/100)**2=1%

probability kinetic energy
will cause major injury
given hitting person

Probability test profiles will
place vehicle in vicinity of
persons

probability Lighting /
Audio is NOT effective

1 minus effectiveness of
Person avoidance
system (probability it
does not work).



Next Steps



Next Steps



- Now to 31 Aug 17
 - Data Reduction
 - Complete Phase 1B – Evaluate monitor performance
 - Generate / refine guidance based on findings – Requirements Development, Design, and Evaluation
- Potential Later Phases
 - Improved FLS risk map development (automated based on real measurements)
 - Incorporation of onboard sensing (less reliance on pre-defined data. i.e. use terrain measurements rather than pre-defined database)
 - Examples: LIDAR, Radar, Optical, Ground Based, Computer Vision, etc.
 - Additional Monitors
 - Examples: Personnel Avoidance (personnel detection and communication of Intent), Air Vehicle Avoidance, Weather Avoidance, etc.
 - Extended range operations
 - Redundancy implications



New vehicle requirements



Take Away



Take Away Points for Developing MMRTA



- Modular architecture with common generic elements
 - Allow specific element development / modification with limited regression testing / verification.
 - Accommodate expandability (E in EVAA)
 - Strive to be as simple as possible while meeting requirements. Avoid unnecessary complexity in code and design of safety systems
 - Configuration files allow for flexibility / expandability
 - CVT JSON file to set list of parameters
 - GCAS / GeoFence config files to set vehicle maneuvers
 - One common data module / interface is essential for integrating multiple monitors (CVT)
 - Each monitor only addresses one risk type
 - Multiple monitors run simultaneously
 - Monitors should not look too far into the future
 - The further out they look the more uncertainty and associated nuisance triggers
- Keep the MMRTA system simple and deterministic, push complexity in to common mission manager systems
 - Each module performs one task or function instead of several (avoid convolution)



Take Away Points for Developing MMRTA



- Minimize computational time to enable responsiveness in the system
 - Solve discrete problems rapidly
 - Use simplified models
 - Implement simple rules / monitors for specific situations
 - Consequence dictates priority
- Monitors should be kept independent (with guidance from requirements on how to interface with the MMRTA architecture)
 - They should not communicate with each other
 - They should each only mitigate one safety risk
 - They should run fast enough to respond faster than the situation can unfold
 - Decision logic for monitor fights (Moral Compass) is crucial to ensure safe function and could easily get complex
 - Consequence method (based on non-dimensionalized dollar impact) simplifies this problem
 - Decisions only need to be made at rates which the combined system (MMRTA and vehicle) can respond to
 - A Monitor acting alone should not be the cause of an incident
 - With multiple Monitors, tackle the highest consequence first, minimizing harm



Take Away Points for Developing MMRTA



- Map data storage (indexing and datum) and usage needs to be clearly defined in the requirements provided to monitor developers.
 - Where in a tile does the lat, lon address actually exist?
 - Posts can create situations where a scan is too small and is between posts and erroneously reports elevations.
 - The horizontal and vertical datum used by the telemetry available from the vehicle should be verified to match the map databases used by the system.
 - If using a terrain database be sure you understand the impact of the source data upon your MMRTA system design: i.e., is terrain data based on same axis system your aircraft uses?
- Decision logic for monitor fights (Moral Compass) design/implementation is the critical element that will establish the perceived behavior of the system to the public
- Requirements Implementation should accommodate ease of verification
 - Flight test enabling interfaces simplified verification testing / troubleshooting of the system
- Think organization (these types of systems can get complex, some software standards are key)
 - Coding style guide and API guides were essential
 - Naming conventions



Take Away Points for Developing MMRTA



- Strongly recommend calculating winds aloft – in majority of cases they are necessary or very helpful to MMRTA.
 - Does system counter winds automatically and stay on desired track/position ... or not ?
 - Essentially a MMRTA system design tradeoff of avoidance clearance vs nuisance avoidance
 - Need driven by aircraft avoidance kinematics coupled with mission operations (i.e, are wind values needed while flying between bldgs.?) Ground trajectory and distance of closest avoidance is function of ratio of vehicle speed to wind speed
- MMRTA should be safety focused not mission focused



Take Away Points for Testing MMRTA



- Evaluate system robustness
 - Plan test profiles that will stimulate the software /avoidances in multiple ways / varying setups
 - Otherwise risk missing unanticipated system responses
 - NASA-Traveler has ~100 test profiles for 3 Monitors (GCAS, GeoFence, & Forced landing System)
 - Testing must include off-nominal conditions & should include testing outside operational envelope
 - Winds > mission ops limits > published op envelope; GPS jamming affects; Reduced sensor-visibility
 - Especially in Cert-matrix cells/subject areas where probability of occurrence or risk to personnel is above a tbd threshold
 - Test matrix density, # of data samples
 - Driven by Probability and Confidence vs. Consequence for total risk for a given system
- Plan for, & Gather data useful for postflight analysis that provides trend data over long periods of time or many flights that may /may not be useful for individual flights
 - Examples: Obtain independent (i.e., not from SUT) 3D position accuracy (not only during test profiles, but also takeoff AND landing to assess drift error over time), number and type of avoidance maneuvers during sorties, time history of winds aloft correlated with aircraft time, distance of closest approach during avoidance maneuvers



Take Away Points for Testing MMRTA



- Don't assume the "advertised" baseline COTS aircraft performance if adding MMRTA to an existing aircraft – verify areas of need/interest
 - Babble testing to ensure no unforeseen complications in data interface between system under test and COTS autopilot
 - Compare a/c envelope against Mission ops demand. Example: If operating in Denver, consider Density-Altitude effects (see supporting briefing slides)
 - Wind effects upon departure susceptibility, avoidance maneuvers, cruise performance
- Don't always assume a/c performance is same across fleet
 - V_{true} & roll response differences between aircraft => Affected avoidance maneuvers and inflight wind calculations => resulted in need for Tail #-specific MMRTA / EVAA parameters in software
- A Simulator /model can be very helpful in MMRTA software development, test matrix planning, and test mission prep, but NOT a valid substitute for MMRTA flight test
 - Req'd fidelity of aircraft kinematics in sim/model depends upon the missions ops
 - Example: if not in constrained flight environment (not closely space multi-obstacles) then High-fidelity of aircraft kinematics is likely NOT necessary



Take Away Points for Testing MMRTA



- In most instances sUAS flight test efficiency possibly more productive and almost as efficient as modeling/simulation
 - Accurate modeling of most sUAS flight dynamics could be more costly than flight test
 - Non-linear, or unpredictable influences are very difficult to model (wind eddys around buildings)
 - Generally, sUAS flights are low cost, quick-turnaround, efficient, safe, etc
- Utilizing a safety pilot (SP) greatly reduces risk during testing
 - Ensure the Pilot-take-control is independent of the SUT (Safety not provided by SUT during evaluation)
 - Verification of Safety Pilot taking and keeping control is crucial (babble / mode switch testing)
- Minimalize the baseline/COTS System reliance upon the System Under Test (SUT) for test safety and impact upon the COTS system performance
 - Otherwise extremely difficult / impossible to isolate the SUT / MMRTA effects from the baseline



Backup



EVAA Consequence Matrix



- Non-dimensional consequence values assigned to each monitor maneuver output based on dollar values associated with loss.

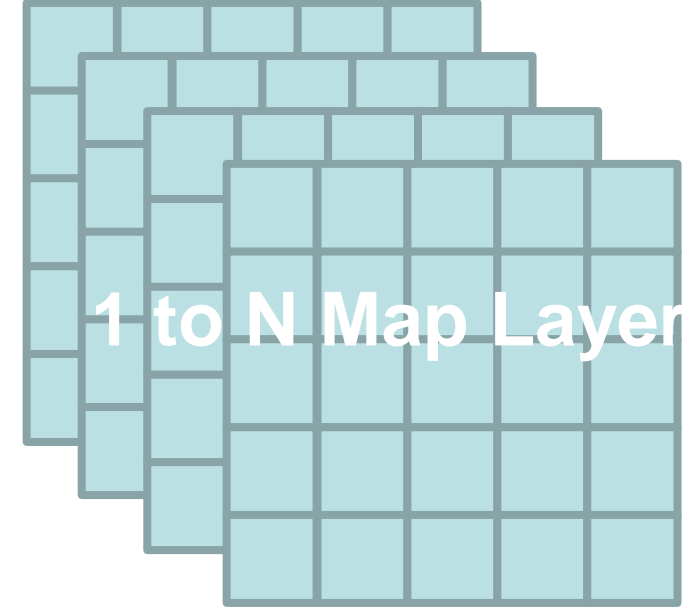
Violation	Consequence	Dollar Value
Waypoint Following	0003	
GeoFence Floor	0148	~\$20
GeoFence Wall	0295	~\$150
GeoFence-Ceiling	0325	~\$200
GCAS-Terrain	0774	~\$2,500
GCAS-Obstacles	1346	~\$12,500
FLS	0982	~\$5,000



Map Manager



- Provides common set of interfaces to geo-referenced raster data.
- Hides details of reading and manipulating geo-referenced raster data from various data formats.
- Base class is a geo-referenced raster cell (GRC).
- A map is composed of a 2-D array of GRC cells. Cells are updated with new data as the vehicle moves.
- The map manager handles multiple layers of maps for different data types.
- Exposes interfaces to access data via shared memory.

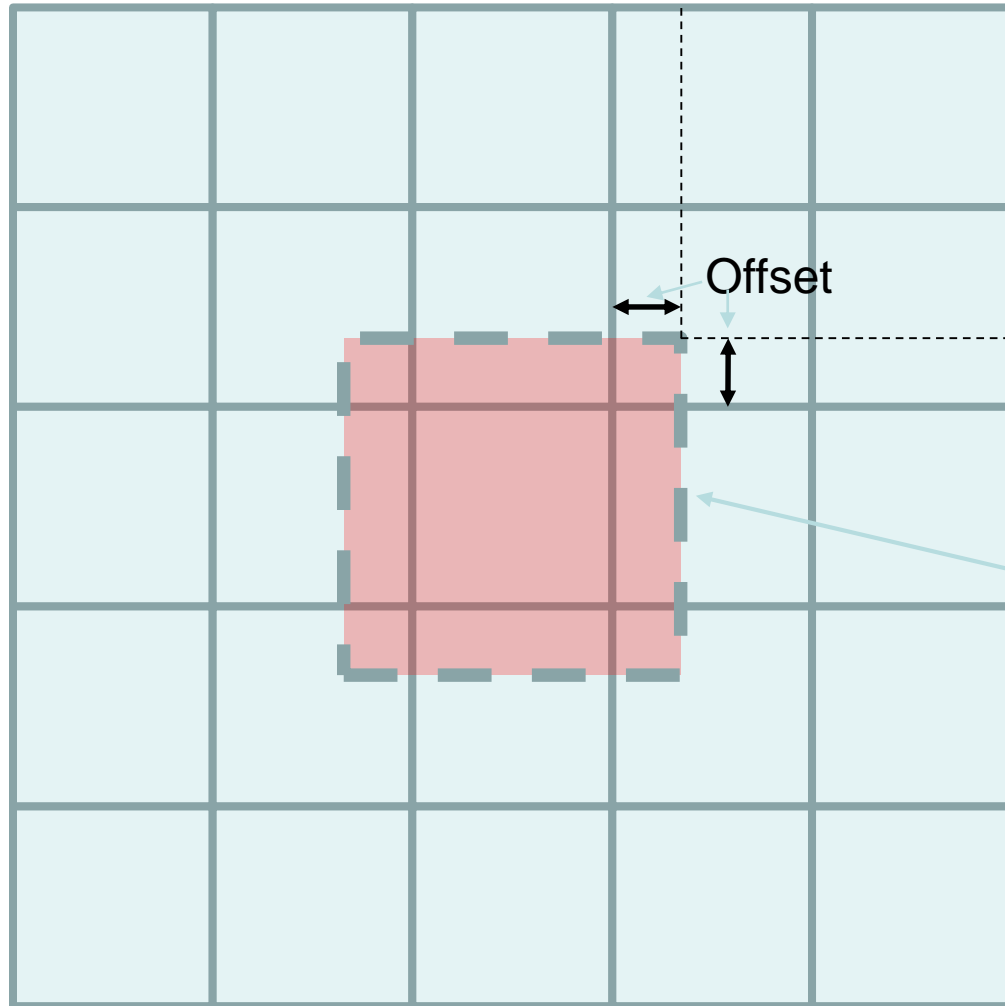




Map Manager Can Read Multiple File Types



- CDTM – Compressed Digital Terrain Map (NASA format)
 - Highly compressed into tiles. Decompressed and re-rasterized.
 - Is not lossless compression. Error limits are imposed on map compression based on end-user requirements.
 - Typically rasterize to same resolution as source data.
- “Feature Data” – project-specific XML format.
 - Easy for user to hand-edit file to add features such as power-lines, buildings, obstacles of various heights and shapes.
- “Format19” – Project-specific 1/9 arc second raster format.
 - Will be used to hold obstacle data generated from LiDAR data.
- DTED – DOD Digital Terrain Elevation Data format
 - Level 2 – 1 arc second resolution, integer elevation (meters)
- NED – National Elevation Data
 - 1/3 arc second resolution, floating point



- Map is considered up-to-date if vehicle is located within the inner-threshold box.
- New rows or columns are loaded in, and old rows or columns of data are 'dropped off' as needed to keep the vehicle location within the inner-threshold box.

$$\text{Offset} = \text{ThresholdFactor} * \text{CellSize}$$

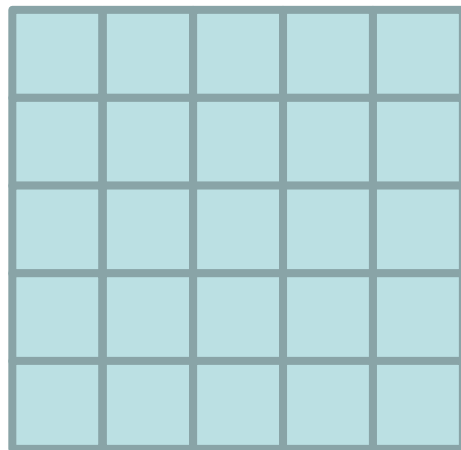
Inner-threshold box



Map Manager Indexing Scheme



2D Array of Indices to GRC Map Cell Array



This 2D array defines the contents of the 5x5 cell moving map.

Using array of indices instead of array of pointers because it was found that pointers did not work well in shared memory between different threads.

1D Array of Indices to 'Next' Cells



This 1D array defines which cells will be used for reading in the next row or column of data, and then be copied into the 5x5 map.

1D Array to temporarily hold indices to unused cells

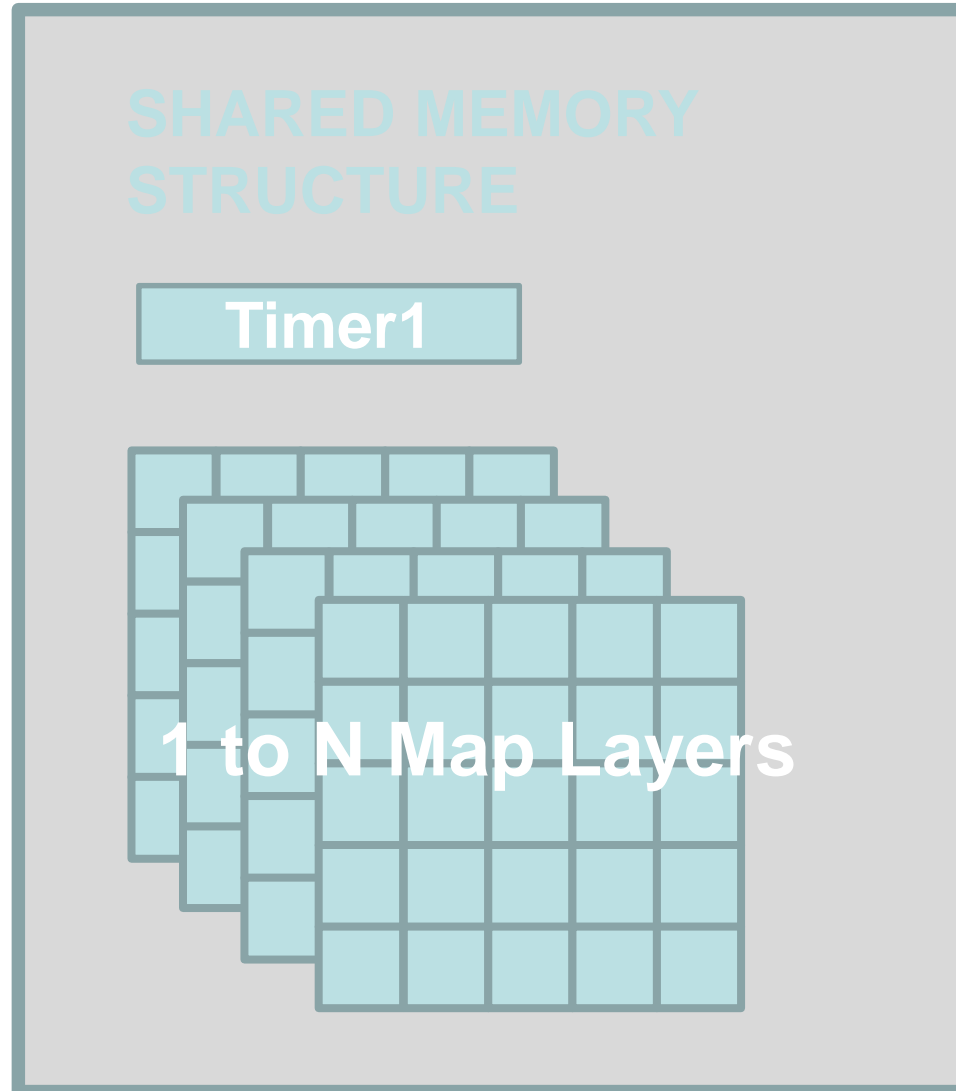


This 1D array holds the indices of the cells being moved off of the 5x5 map, while the other cells are being shifted. After the cells are shifted and the contents of this array are copied to the 'Next' array, the values in this array are set to a value that indicates the contained indices are invalid.

- 0
- 1
- 2
- 3
- 4
- 5
- 6
- 7
- 8
- 9
- 10
- 11
- 12
- 13
- 14
- 15
- 16
- 17
- 18
- 19
- 20
- 21
- 22
- 23
- 24
- 25
- 26
- 27
- 28
- 29



Map Manager Shared Memory



Map Control Interfaces:

- **ConnectAsOwner()**
- **ConnectAsUser()**
- **Initialize(mapSelect, ...)**
- **bMapsAreInitialized()**
- **SetMapCenter()**
- **UpdateMap()**

Data Access Interfaces:

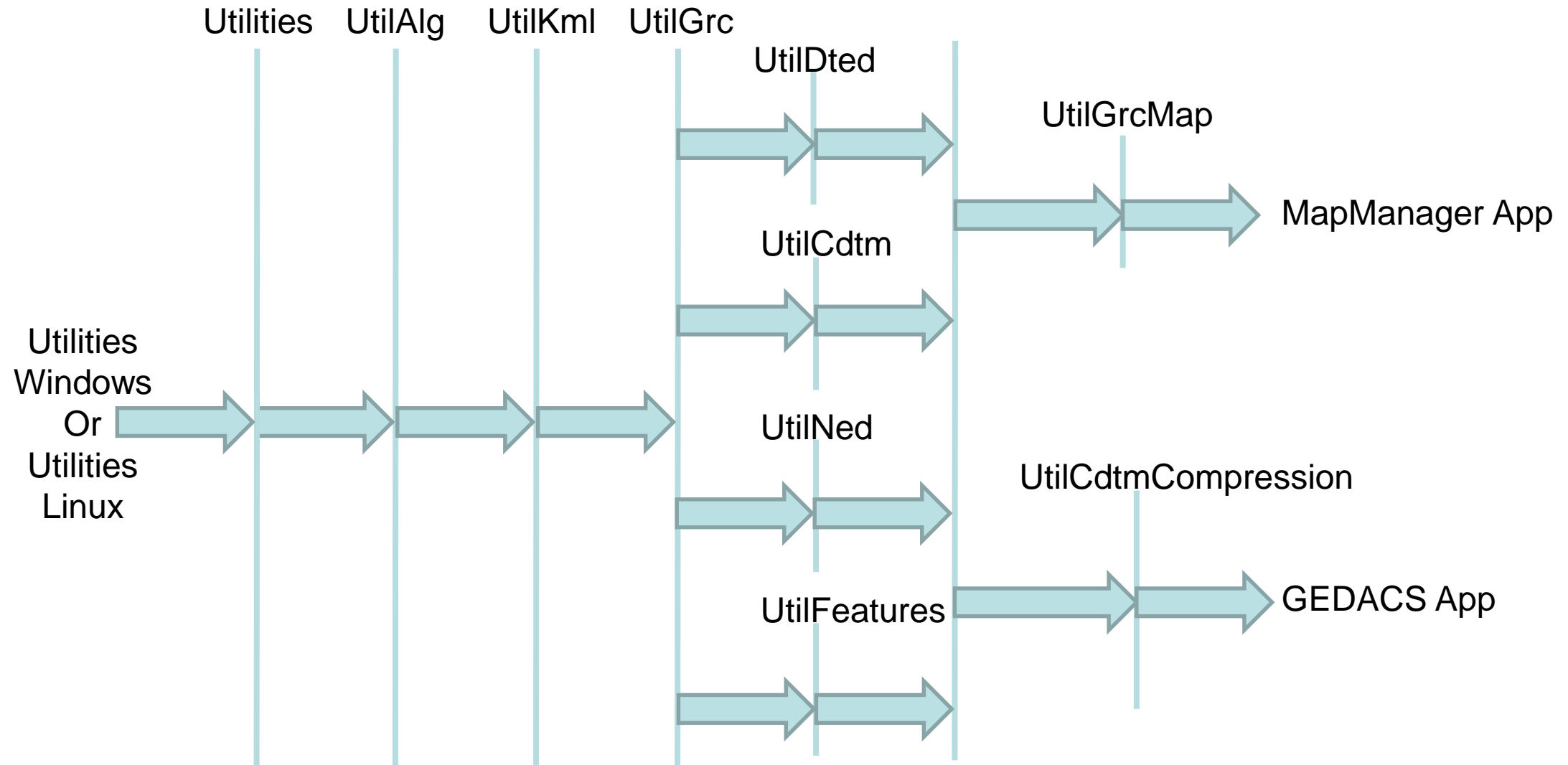
- **GetElevAtPos()**
- **GetHighestPointWithinCircle()**
- **GetListOfPtsWithinCircle()**

TimerInterfaces:

- **ResetTimer1()**
- **GetTimer1TimeSinceReset_sec()**



Map Manager Library Dependencies



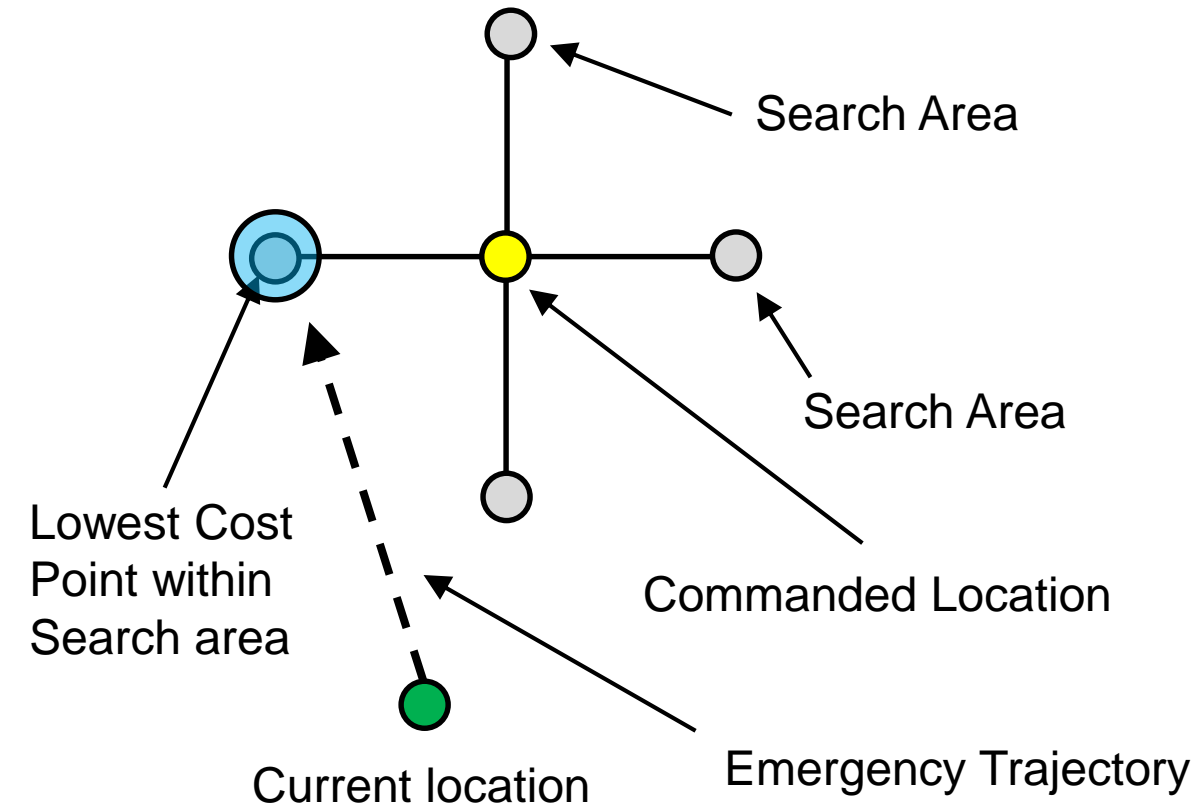


Backup

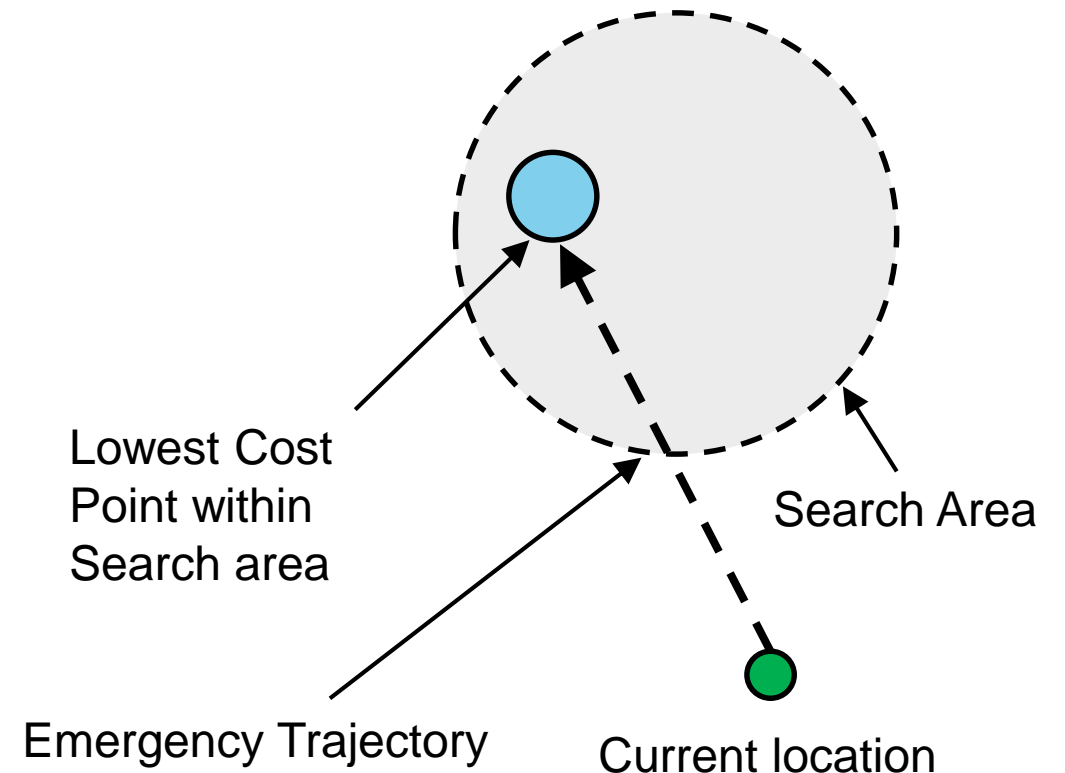


The Traveler FLS (Forced Landing System) provides an emergency landing location and autopilot commands to minimize injury and property damage during an inflight emergency. Current implementation has two modes that are triggered by external command

Land Here



Land Immediately





Velodyne VLP-16 LiDAR

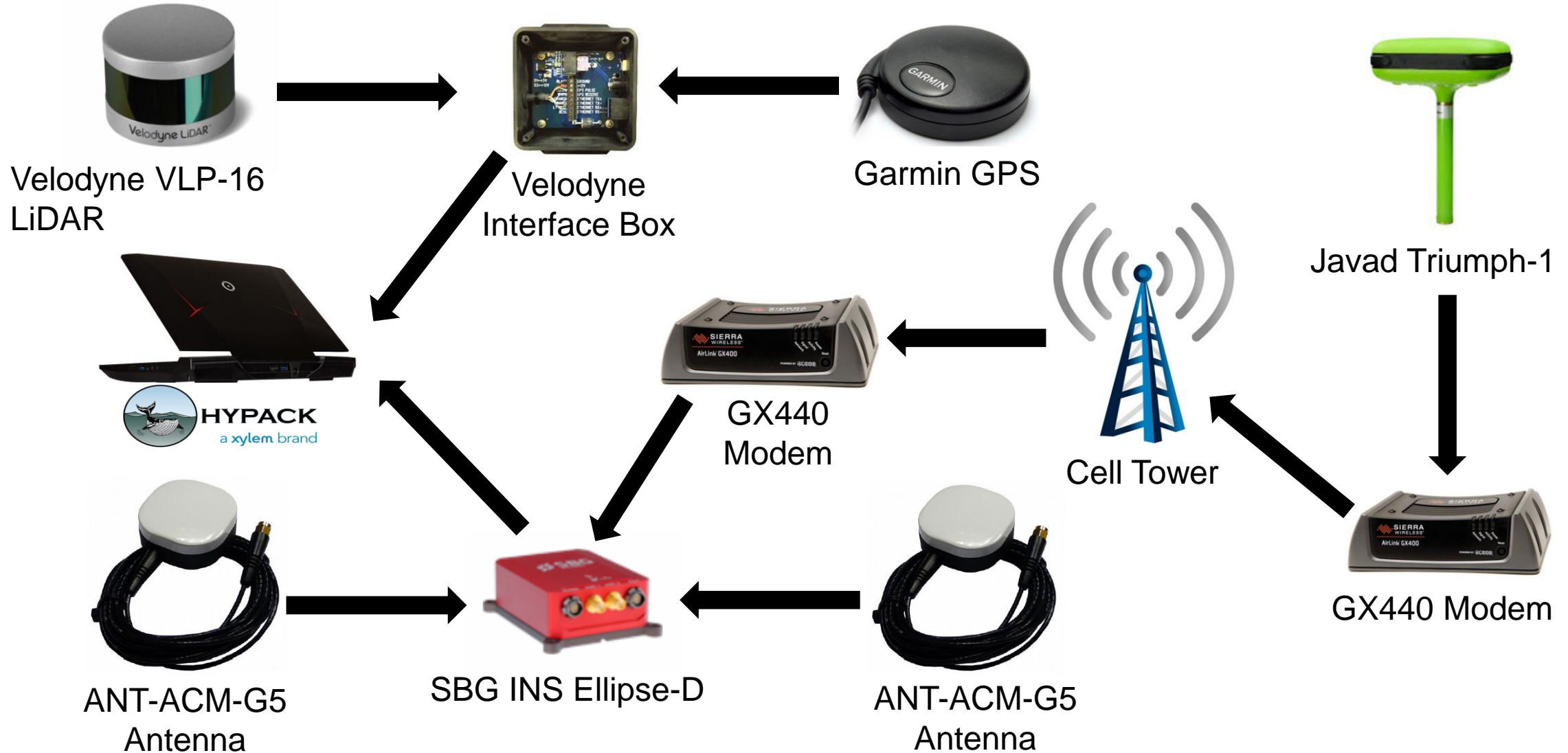


- 16 Lasers
- 360° Horizontal FOV
- $\pm 15^\circ$ Vertical FOV
- 300,000 Points per Second
- 100m Range
- Will be used for
 - Feature/Risk map generation
 - Active Sensing





LiDAR System



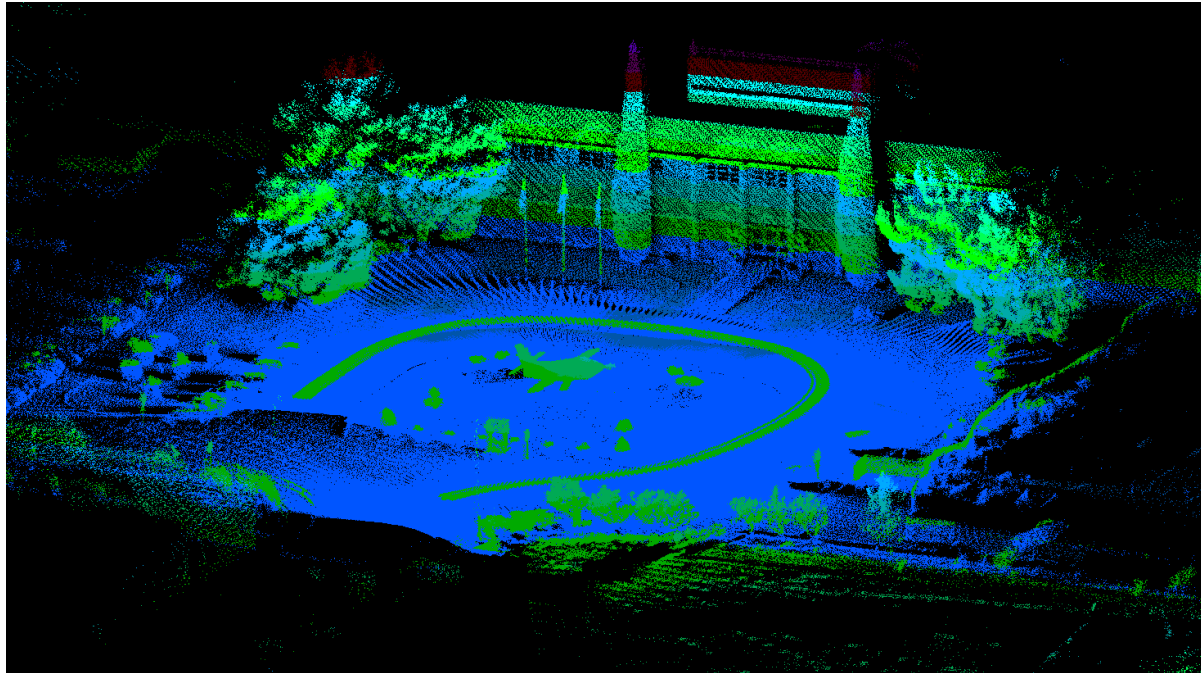


LiDAR System Setup





LiDAR System Sample Data





LiDAR System Sample Data

